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# COLLOCATION FLUTTER ANALYSIS

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VOLUME I.

## GENERAL PROGRAM DESCRIPTION

APRIL 1969



MISSILE SYSTEMS DIVISION

HUGHES

HUGHES AIRCRAFT COMPANY

SEP 25 1969

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COFA

COLLOCATION FLUTTER ANALYSIS STUDY

VOLUME I

THE COFA PROGRAM

Prepared by the Dynamics and Environment  
Section Personnel, Hughes Aircraft Company  
Under Contract Number 0019-68-C-0247

APRIL 1969

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ABSTRACT

This study covers the development of a set of computer programs to perform flutter analysis by the collocation method. While this method has been known for some time, only recently have advances in computer technology made the method technically and economically feasible. The ingredients of a collocation flutter analysis are (1) a flexibility matrix, (2) aerodynamic influence coefficient matrix, and (3) an eigenvalue solution. This study is presented in four volumes. Volume I contains a general program discussion. Volume II contains the program FLUENC which calculates the flexibility matrix. Volume III contains a set of three programs to calculate aerodynamic influence coefficients for subsonic, transonic, and supersonic flight regimes. Volume IV contains the program COFA which sets up and solves the flutter eigenvalue matrix.

## FOREWORD

This report covers the research conducted by the Missile Systems Division of Hughes Aircraft Company, Canoga Park, California, under Contract Number 00019-68-C-0247.

This work was performed to develop a comprehensive set of computer programs that perform flutter analysis using the collocation method. A set of three computer programs to calculate unsteady aerodynamic influence coefficients for subsonic, transonic, and supersonic lifting surfaces with interaction between the wing and a downstream control surface lying in the wake of the wing are presented. A program to calculate structural influence coefficients is presented, and a program to set up and solve the flutter eigen matrix is presented.

The Program Manager for Hughes Aircraft Company was R. J. Oedy. Dr. V. Weingarten assisted in the development of the structural analysis program, and Dr. W. P. Rodden assisted in the development of the aerodynamic programs and flutter analysis programs.

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## PART I - INTRODUCTION

Collocation is a term used to describe a mathematical procedure for (among other things) solving integral equations. The word literally means to co-locate, i.e., an approximate solution is co-located to satisfy the equation at a specified discrete number of locations in the space (the collocation points). When the integral equation for vibration analysis is solved by the collocation method it is generally referred to as the lumped-parameter method since the properties of the continuum are assumed to be concentrated ("lumped") at the collocation points. The collocation method leads to a formulation of the flutter problem similar to that of the vibration problem where, in addition to the structural influence coefficients (SIC's) and mass matrix, aerodynamic influence coefficients (AIC's) are required.

The vibration problem can be large if many degrees of freedom are necessary to model the vibrating system, and the calculation of the modes and frequencies requires solving a large eigenvalue problem. However, since both the SIC's and mass matrix are real symmetric matrices, special techniques can be utilized to find the modes and frequencies rapidly (and, hence, economically). The corresponding formulation of the flutter problem does not possess any special properties that permit convenient determination of the modes, frequencies, and flutter velocities. For this reason a series or modal solution has traditionally been employed to reduce the size of the flutter problem and the corresponding computing time.

Present day computer capacities and speeds, and advances in eigenvalue methods however, no longer limit flutter analysis to modal methods and the advantages of a direct solution can be realized. One advantage is that questions of convergence of the modal solution are avoided. Another advantage is that the flexibility, mass and aerodynamic data are kept separate so that a change in any one set can be made independently of the other two. Finally, the flutter analysis can be done without performing a vibration analysis. This is a particular advantage when variations in mass or stiffness distribution must be investigated. Examples include control surface mass balance optimization studies or weight reduction due to burning fuel.

## PART II - GENERAL DISCUSSION OF FLUTTER

### A. The Mechanism of Flutter

Flutter is a self-excited vibration of a flexible body in a fluid stream. The flutter speed is defined as the minimum speed at which the aeroelastic system will sustain a neutrally stable oscillation. At lower speeds the motion will be damped, and at higher speeds, or at least in a range of higher speeds, the motion will be divergent. Usually only a small increase beyond the flutter speed is required to produce a rapid divergence of such violence that a complete structural failure results in only a few cycles of motion.

At the flutter speed, a critical phasing between the motion and the loading permits extraction of an amount of energy from the air stream equal to that dissipated by internal damping during each cycle and thereby sustains a neutrally stable periodic motion. The simplest type of flutter occurs when a simple motion induces an aerodynamic force having a component in the direction of the motion and in phase with the velocity. This is described as a case of negative aerodynamic damping or single degree of freedom flutter. The term classical flutter is used to denote the more complicated instability that typically arises from a critical coupling of two or more modes of motion, each of which is usually stable by itself. It is the classical flutter mechanism with which we are primarily concerned because its generality includes the single degree of freedom instability as a special case.

### B. The Ingredients for Flutter Analysis

Since flutter is a dynamic aeroelastic instability the forces involved are structural, aerodynamic, and inertial. In the present lumped-parameter or collocation formulation these appear as matrices of structural influence coefficients (SIC's), aerodynamic influence coefficients (AIC's), and masses. The combination of the various matrices to formulate the flutter stability problem leads to a non-Hermitian eigenvalue problem. Since this is the most general type of eigenvalue problem, methods for its solution are quite limited. The method that has been employed is the power method, or as it is sometimes called, the matrix iteration method.

## PART III - DISCUSSION OF COFA PROGRAM

### A. The Flutter Analysis

#### 1. The Technical Approach -

The collocation method of flutter analysis had been known for many years but it was not demonstrated to be computationally feasible until 1956 in Reference 1. The method was extended to include an arbitrary number of rigid body degrees of freedom in Reference 2. A complete discussion of the general flutter analysis is given in the Flutter Analysis Report. It is sufficient for the introductory purposes of this report to consider the formulation of the flutter problem for a restrained vehicle.

The basic equation is the relationship between the deflections  $\{h\}$  to the forces  $\{F\}$  through the structural influence coefficients (SIC's)  $[a]$ .

$$\{h\} = [a] \{F\} \quad (3.1)$$

The force matrix must include both the inertial and aerodynamic forces.

$$\{F\} = -[M] \{\ddot{h}\} + \{F_a\} \quad (3.2)$$

where  $[M]$  is the mass matrix corresponding to the chosen set of control points. We define a complex matrix of oscillatory aerodynamic influence coefficients (AIC's),  $[C_h]$ , by the equation

$$\{F_a\} = \rho \omega^2 b_r^2 s [C_h] \{h\} \quad (3.3)$$

where  $\rho$  is the density,  $\omega$  is the frequency,  $b_r$  is the reference semi-chord, and  $s$  is the reference semi-span. Combining Eqs. (3.2) and (3.3) for harmonic motion, we find the total force to be

$$\{F\} = \omega^2 \left( [M] + \rho b_r^2 s [C_h] \right) \{h\} \quad (3.4)$$

If we substitute Eq. (3.4) into Eq. (3.1) and divide the static SIC's by  $(1 + ig)$  where  $g$  is the artificial structural damping necessary to sustain the assumed harmonic motion, we obtain the matrix equation for flutter

$$\{h\} = \frac{\omega^2}{1 + ig} [a] \left( [M] + \rho b_r^2 s [C_h] \right) \{h\} \quad (3.5)$$

Eq.(3.5) can be solved for the complex mode shape,  $\{h\}$ , and the complex eigenvalue,  $(1 + ig)/\omega^2$ , by complex matrix iteration (the power method). From the eigenvalue

$$\lambda = \lambda_R + i\lambda_I \quad (3.6a)$$

$$= (1 + ig)/\omega^2 \quad (3.6b)$$

we obtain the flutter frequency

$$\omega = 1/\sqrt{\lambda_R} \quad (3.7)$$

the required artificial damping

$$g = \lambda_I / \lambda_R \quad (3.8)$$

and, since the AIC's required the assumption of a reference reduced frequency,  $k_r = \omega b_r / U$ , for its calculation, we find the velocity

$$U = \omega b_r / k_r \quad (3.9)$$

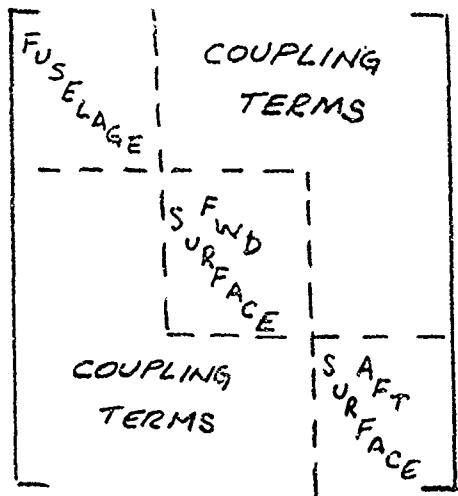
From a series of solutions for different reduced frequencies, a required damping velocity curve can be constructed for a specific altitude. The flutter velocity corresponds to the lowest velocity at which the required damping is equal to the actual structural damping of the system.

## 2. Program Capabilities

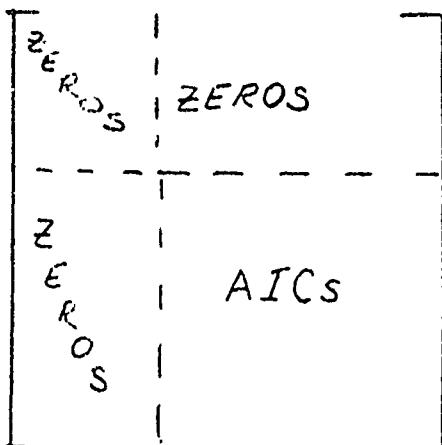
The COFA program has the limitations of analyzing a system with a maximum of 49 collocation stations. In the use of these 49 stations, there is a great deal of flexibility. Entire vehicles may be analyzed as single entities or as conglomerates of components with single or tandem aerodynamic surfaces. Both may be analyzed. When the components are used to synthesize a structure, the attachment between the components must be statically determinant. The components may be either flexible or rigid. The coupling of flexible components is demonstrated in the example problem of this volume. The coupling of flexible to rigid components is demonstrated in the example problems of Volume IV. The systems analyzed may be cantilevered or free in space. Up to six rigid body modes may be used in the analysis of systems free in space. In addition, there is the capability of using a free-free boundary condition built into the program for freeing cantilevered or fixed systems.

There are two classes of tandem surface problems; they are aerodynamically non-interfering surfaces and aerodynamically interfering surfaces. For non-interfering surfaces the structural system may be analyzed as a single entity or a conglomerate of components. The most usual way is to use the component synthesis technique. When the component technique is used, the AIC, SIC, and mass matrices are partitioned with a partition corresponding to each structural component. This results in dealing with smaller order matrices, and the analysis is straightforward as presented in Volume IV. The only precaution is that the mass, flexibility, and aerodynamic matrix partitions be entered serially into the computer program in an identical manner, so that the information is properly located in the overall dynamical matrix, i.e. fuselage first; forward surface second; and aft surface third; etc. For interfering surfaces, the analysis must be conducted as a single structural entity; for interfering surfaces the following procedure must be followed:

a. When calculating the structural influence coefficients the fuselage characteristics must be entered first, followed by the forward surface then the aft surface. This results in a flexibility matrix of the following form



b. The COFA aerodynamic programs do not calculate aerodynamic forces for the fuselage; thus, in order to enter the aerodynamics properly into the program, the aerodynamic matrix must be banded by zeros as follows



If fuselage aerodynamics are obtained from a different source, they may be properly entered into the analysis in the upper left partition. The off diagonal partitions are for wing-fuselage interference effects. The details of using the COFA Program may be found in Volume IV.

#### B. The Calculation of Structural Influence Coefficients

##### 1. The Technical Approach -

There are many methods by which structural influence coefficients may be calculated. The SIC's calculated for this program use the direct stiffness method. The direct stiffness method was chosen as it is readily adapted to highly redundant structures and is especially suited to the use of high speed digital computers. The structure is regarded as an assemblage of beams and plates. The stiffness matrix for the entire structure is computed by the simple summation of the stiffness matrices of the elements of the structure. Finally, the matrix of SIC's is obtained by the inversion of the stiffness matrix.

For a given idealized structure, the analysis of stresses and deflections due to a given system of loads is a purely mathematical problem. Two conditions must be satisfied in the analysis: 1) the forces developed in the members must be in equilibrium and, 2) the deformation of the members must be compatible - i.e., consistent with each other and the boundary conditions. In addition, the forces and deflections in each member must be related in accordance with the stress-strain relationship assumed for the material. These conditions are satisfied by the direct stiffness method in the following way. The displacements of the joints of the structure are considered as unknown quantities. Thus, an infinite number of systems of mutually compatible deformations in the members are possible; the correct pattern of displacements is the one for which the equations of equilibrium are satisfied. The concept of static determinateness or indeterminateness is irrelevant when the analysis is considered from this viewpoint. This approach is the basis for many relaxation type analysis (such as moment distribution) and is used extensively in the analysis of complex aircraft structures.

## 2. Program Capabilities -

The program FLUENC calculates structural influence coefficients for systems with up to 50 mass points. The mass points may be connected by any combination of beams or plates. Each member may have its own material property. The program also calculates the natural frequencies and mode shapes of the structure. The program can calculate the modes and frequencies for structures fixed or free in space. Caution should be exercised not to use the flexibility matrix of structures free in space as the flexibility matrix of such a structure cannot be directly calculated. The flexibility matrix of a free structure can be obtained by 1) arbitrarily fixing the structure, 2) calculating the flexibility matrix for the fixed structure, 3) enter the flexibility matrix of the fixed structure into the COFA program and use the free-free boundary condition to obtain the flexibility matrix of the free-free structure. The details of using the program FLUENC may be found in Volume II.

### C. The Unsteady Aerodynamic Lifting Surface Theories

#### 1. The Subsonic Kernel Function Procedure

##### a) The Technical Approach -

The linearized equation for the perturbation potential  $\phi$  for a body immersed in a compressible, inviscid, perfect gas, when we assume the flow to be isentropic and irrotational, is

$$(1 - M^2) \frac{\partial^2 \phi}{\partial x^2} + \frac{\partial^2 \phi}{\partial y^2} + \frac{\partial^2 \phi}{\partial z^2} = \frac{1}{a_0^2} \frac{\partial^2 \phi}{\partial t^2} + \frac{2M}{a_0} \frac{\partial^2 \phi}{\partial x \partial t} \quad (3.10)$$

in which  $M$  is the Mach number,  $a_0$  is the ambient speed of sound, and  $x, y, z$  are the Cartesian coordinates of the flow field. The linearized perturbation pressure coefficient is related to the potential through

$$c_p = - \frac{2}{U^2} \left( \frac{U \partial \phi}{\partial x} + \frac{\partial \phi}{\partial t} \right) \quad (3.11)$$

where  $V$  is the free-stream velocity. The problem in lifting surface theory is to find the difference in the pressure coefficients between the upper and lower surfaces, i.e., the lifting pressure coefficient. The evaluation of Eq.(3.11) requires the solution of Eq. (3.10). The differential equation (3.10) must be solved subject to certain boundary conditions. There can be no flow through the surface, i.e., at the surface the normal velocity of the fluid must equal the normal velocity of the surface (the downwash velocity). Additional conditions at subsonic speeds require that the pressure must remain finite at the trailing edge (the Kutta condition) and the potential must vanish at an infinite distance from the surface. The additional conditions at supersonic speeds are that no disturbance can propagate upstream of the leading edge, and that the disturbances must propagate from the surface as outgoing waves. At transonic speeds a combination of the subsonic and supersonic boundary conditions are required.

The foregoing general review of lifting surface theory leads us to consider specific Mach number regimes. The first numerical method for solving the subsonic pressure distribution for planar lifting surfaces undergoing simple harmonic motion was developed by Watkins, Woolston, and Cunningham (Reference 3). The approach taken was to recast the differential equation for the velocity potential into an integral equation relating the surface downwash to the lifting pressure coefficient (the acceleration potential). The solution was carried out by assuming a series form for the pressure coefficient based on known exact solutions, e.g., the two-dimensional solution for chordwise pressure, and the elliptical spanwise loading on an elliptical wing. From this series of pressure modes the problem reduced to solving for the amplitude of each pressure mode. A collocation method was used in which the downwash was matched at a discrete number of points. The numerical difficulties are numerous since the integral equation is singular and the many integrals of the pressure modes require extreme accuracy. Since the publication of Reference 3, many investigators have refined the procedures, notably Hsu (Reference 4), Vivian and Andrew (Reference 5), and Rowe (Reference 6). The present study is an extension of References 7 and 8 to obtain AIC's, thereby eliminating the limitations introduced by the requirements for polynomial surface fitting in References 7 and 8. The details of the AIC's and the numerical problems of the subsonic lifting surface theory are discussed in the Unsteady Aerodynamics Report, Part III and Part IV, Section A, respectively.

b) Program Capabilities -

The Subsonic AIC program calculates AIC's for systems with up to 40 collocation stations. The systems analyzed may be a single surface or tandem surfaces. Up to 20 collocation stations per surface may be used. When analyzing single surfaces, a dummy trailing surface must be input and isolated from the leading surface. The following single surface planforms may be analyzed: rectangle, trapezoidal, delta, cropped delta and cropped trapezoid. The trailing edge of the surface must be straight and normal to the direction aerodynamic flow. When analyzing tandem surfaces, the forward surface may be anyone of the single surfaces previously defined; the aft surface must be a rectangular surface with the spanwise dimension equal to the spanwise dimension of the wing.

The program automatically sets the number of chordwise and spanwise pressure series terms equal to the number of chordwise and spanwise collocation station. This maximizes the number of pressure series terms used and increases computation accuracy. For best accuracy a minimum of three chordwise collocation stations should be used. The number of kernel function chordwise integration stations is a variable in the program, and is a function of the number of chordwise collocation stations. (The number of integration stations equals factor times the number of chordwise collocation stations.) The factor, NI<sub>0</sub>NCX, which is input data, should be such that the minimum number of integration stations is 15. The maximum number of integration stations per surface is 40.

When analyzing single surfaces, the dummy surface is required to have two chordwise collocation stations. The number of spanwise collocation stations on the dummy surface is established internally in the program and is set equal to the number of spanwise collocation stations used in the analysis of the single surface. The details of using the Subsonic AIC program may be found in Volume III.

## 2. The Transonic Box Method

### a) The Technical Approach -

The basic development of the transonic box method was made by Rodemich and Andrew (Reference 9). References 7 and 8 include extensions of Reference 9 to interfering wing-tail configurations. The basic formulation of the transonic potential equation, pressure coefficient, and boundary conditions has been included in the introductory discussion of the technical approach to the subsonic problem above in Section A, and the details of the AIC's and the numerical problems of the transonic box method are discussed in the Unsteady Aerodynamics Report, Part III, and Part V, Section A, respectively. For the purposes of this introductory summary it is sufficient to outline the basic procedure.

The transonic box method is designed to calculate the unsteady potentials from which the pressure distributions may be obtained for arbitrary modes of surface motion. The method used was suggested by the successes of the supersonic box methods (References 10 and 11). The potential is generated by a doublet distribution rather than by a source distribution because the latter method would involve diaphragm regions of infinite extent, whereas the doublet distribution is confined to the wing and its wake. As with the subsonic problem, the differential equation is rewritten as an integral equation. The integral equation is approximated numerically by a matrix equation so that the basic step in the box method is the solution of the system of simultaneous equations which determine a set of values of potential on the surface from a corresponding array of upwash values. The solution procedure obtains the velocity potential over the surface one spanwise row of boxes at a time until the trailing edge row is completed. The numerical complexity is not increased, however, by a large number of box rows over the configuration because the influence coming from more than 15 rows away has been found to be negligible. The results are valid for high reduced frequency  $k$ , such that  $k \gg |M - 1|$  where  $M$  is the Mach number.

b) Program Capabilities -

The Transonic AIC program calculates AIC's for systems with up to 40 collocation stations. The systems analyzed may be a single surface or tandem surfaces. Up to 20 collocation stations per surface may be used. The following single surface planforms may be analyzed: rectangular, trapezoidal, delta, cropped delta, and cropped trapezoid. It is not necessary to enter a second surface into the program when analyzing a single surface. The trailing edge of the surface must be straight and normal to the zero-dynamic flow direction. When analyzing tandem surfaces, the forward surface may be any one of the single surfaces previously defined; the aft surface must be a rectangular surface with the spanwise dimension equal to the spanwise dimension of the wing.

The maximum allowable number of Mach boxes is 45. The layout of the boxes on the surface is determined in the program, and is a function of the number of chordwise boxes. The Mach boxes are square, and can be located on the surface by placing the center of one box at the intersection of the root chord and the trailing edge of the forward surface. The length of the side of a Mach box is approximately equal to the root chord length divided by the number of chordwise boxes chosen. It is not necessary to layout the Mach boxes, as this is done by the program, but this knowledge is helpful in choosing the optimum number of chordwise Mach boxes. In general, the accuracy increases as the number of Mach boxes increase. Therefore, the maximum number of Mach boxes should be used. The details of the Transonic AIC program may be found in Volume III.

### 3. The Supersonic Box Method

#### a) The Technical Approach -

The basic work on the supersonic box method was done by Pines, Dugundji, and Neuringer (Reference 10) and by Zartarian and Hsu (Reference 11). Further refinements were made by Moore and Andrew (Reference 12). References 7 and 8 also include extensions of Reference 12 to interfering wing-tail configurations. The fundamentals of the potential equation have been presented above in the introductory discussion of the technical approach to the subsonic problem in Section A, and the details of the AIC's and the numerical problems of the supersonic box method are discussed in the Unsteady Aerodynamics Report, Part III and Part VI, Section A, respectively. We merely outline the basic procedure in this introductory summary.

Pines, et al., (Reference 10) published the first source superposition method to approximate successfully the aerodynamic forces on an oscillating thin planar surface in supersonic flow. They employed Evvard's diaphragm concept (Reference 13) to handle subsonic leading edges and overlaid the surface and diaphragm with a grid of square boxes. For purposes of calculating pressures, they assumed that the source strength over the area of each box is a constant value which satisfies the condition of tangential flow at the center of the box. The Mach-box procedure is basically the same as the method of Pines et al., differing only in that the surface and diaphragm is overlaid with a grid of rectangular boxes, the diagonals of which are parallel to Mach lines. As in the Subsonic and Transonic cases the potential equation is written as an integral equation, this time relating the downwash to the source strengths, and approximated by a matrix equation. The numerical difficulties are primary ones of computer logic since the zones of influence of a given Mach box is limited to the region within the aft Mach lines. The matrix formulation leads to a partitioned form since there are two boundary conditions to be matched. The first boundary condition is the downwash on the surface, and the second is that there be no pressure difference off of the surface in the diaphragm regions. The zero pressure conditions leads to a relationship between the diaphragm potentials and the surface potentials, and the surface downwash condition then leads to the surface potentials. The surface pressures then follow from the surface potentials.

b) Program Capabilities -

The Supersonic AIC program calculates AIC's for systems with up to 40 collocation stations. The systems analyzed may be a single surface or tandem surfaces. Up to 20 collocation stations per surface may be used. The following single surface planforms may be analyzed: rectangular, trapezoidal, delta, cropped delta, and cropped trapezoid. It is not necessary to enter a second surface into the program when analyzing a single surface. The trailing edge of the surface must be straight and normal to the direction of flow. When analyzing tandem surfaces, the forward surface may be any one of the single surfaces previously defined; the aft surface must be a rectangular surface with the spanwise dimension equal to the spanwise dimension of the wing.

The maximum allowable number of Mach boxes is 45. The layout of the boxes on the surface is determined in the program, and is a function of the number of chordwise boxes and the Mach number. The Mach boxes are rectangular with a length in the chordwise direction equal to approximately the root chord divided by the number of chordwise Mach boxes, and a length in the spanwise direction equal to the length in the chordwise direction divided by the square root of the Mach number squared minus one. The Mach boxes can be located on the surface by placing the center of one box at the intersection of the root chord and the trailing edge of the wing. It is not necessary to layout the Mach boxes, as this is done by the program, but this knowledge is useful in choosing the correct number of chordwise Mach boxes. In general, the accuracy increases as the number of Mach boxes increase. Therefore, the maximum number of Mach boxes should be used. The details of the Supersonic AIC program may be found in Volume III.

#### PART IV - SAMPLE PROBLEM

As an example problem, a typical missile with tandem aerodynamic surfaces is used. The tandem surfaces are in close proximity such that aerodynamic interference between surfaces occurs. Thus, the structure must be modeled as one entity. The analysis will be performed for the missile free in space with two rigid body modes (pitching and plunging); thus, the free boundary condition of the COFA program will be employed.

The entire missile is flexible. For a missile free in space, two basic modes of flutter can exist. They are symmetrical and anti-symmetrical flutter; these modes of flutter are demonstrated in Figure 4.1. An analysis for symmetrical flutter is presented. Due to symmetry, only one half of the structure needs to be included in the analysis. A sketch of the missile to be analyzed is shown in Figure 4.2.

The first step in the analysis is to develop a structural idealization (mathematical model) that is consistent with the program FLUENC. The mathematical model established for this example is shown in Figure 4.3. The structure was arbitrarily fixed in translation at mass points 5 and 7, as a flexibility matrix for a structure free in space can not be directly calculated. These points will be released later using the free-free boundary condition of the COFA program. The next step is to run the program FLUENC to generate the flexibility matrix. The computer program FLUENC normally yields a reduced flexibility matrix: i.e. the rows and columns pertaining to the fixed points are eliminated. This is satisfactory when the constrained points are included in a rigid component or a fixed reference point; however, when the constrained points are part of a flexible structure or component this is not satisfactory. A full flexibility matrix is required in order that the entire inertia of the structure may be included. As a consequence, the option to punch the expanded flexibility matrix for the structure was necessary. This option inserts rows and columns of zeros in the flexibility matrix corresponding to the constrained mass points. Next a set of flight parameters were chosen and the aerodynamics program was run. For this example the subsonic AIC program was used, the flight parameters used are

Mach number = .5  
Reduced Frequency = 1.0, 1.5, 2.0  
Altitude = sea level

The computer output for the subsonic AIC program is shown on pages 38 through 42 . As the AIC programs only establish AIC's for the aerodynamic surfaces, the output from the computer program must be banded by the proper number of zeros, corresponding to the number of mass points on the fuselage. The flexibility matrix, the modified AIC matrix, the proper rigid body modal matrices corresponding to symmetrical flutter, and a rigid body mass matrix corresponding to the arbitrarily fixed points in the FLUENC analysis, are entered into the COFA program. The output from the COFA program is shown on pages 61 through 65 . The pertinent flutter information is shown on page 65 .

The data shown is for the specific flight parameters entered. For the output data to be valid, the flutter velocity must equal the assumed input velocity. When this occurs, the damping value yielded is the damping required at that velocity and is a point on the flutter boundary. Thus, determining a point on the stability boundary is an iterative procedure based upon varying the reduced frequency for a given set of flight parameters until the flutter velocity is approximately equal to the input velocity. In general, this occurs at a different reduced frequency for each flutter mode; thus, for one set of flight parameters, there are as many iterative analyses required as there are modes requested in the analysis. The above analysis determines one point on the flutter boundary. To determine the entire flutter boundary, the above multiple iterative procedure must be repeated for several sets of flight parameters. A typical "V-g" curve is shown in Figure 4.4 for a given altitude. Using a set of these curves, for several altitudes, a flutter boundary of the type specified in MIL-A-8870 or MIL-M-8856 may be constructed.

When solving an antisymmetrical flutter problem the technique is identical to that of the symmetrical problem with the exception of the fuselage structural data input for calculating the flexibility matrix. For the antisymmetrical case, the fuselage torsional stiffness and polar moments of inertia rather than the bending stiffnesses and translational masses are required. In addition, the fuselage mass points must be restrained in translation, allowing only rotation of the fuselage. The mass and stiffness data for the aerodynamic surfaces is identical to that for the symmetrical analysis.

As noted above, the analyses are an iterative procedure. Thus, whether an analysis is valid or not is an arbitrary decision, as the output velocity will never exactly be the input velocity. Normally when the output velocity is within five percent of the input velocity, the results are considered valid. For the example problem, an input velocity of 320 kts (Mach .5) was used. The output velocity for the second mode, using a  $1/k_r = .660$ , was 308 kts (page 65). This normally would be considered a valid point on the flutter boundary for the second mode. Scanning the results on page 65 it can be seen that the valid data point for the first mode requires a  $1/k_r$  value between 1.00 and .666; also for the third mode a value between .666 and .500. For modes 4, 5 and 6, the  $1/k_r$  value must be greater than 2.0. It is noted that the function of flutter velocity versus  $1/k_r$  is not generally linear; thus, interpolation will not necessarily yield the correct  $1/k_r$  to use in subsequent iterations.

FIGURE 4/1  
TYPICAL MODES OF FLUTTER

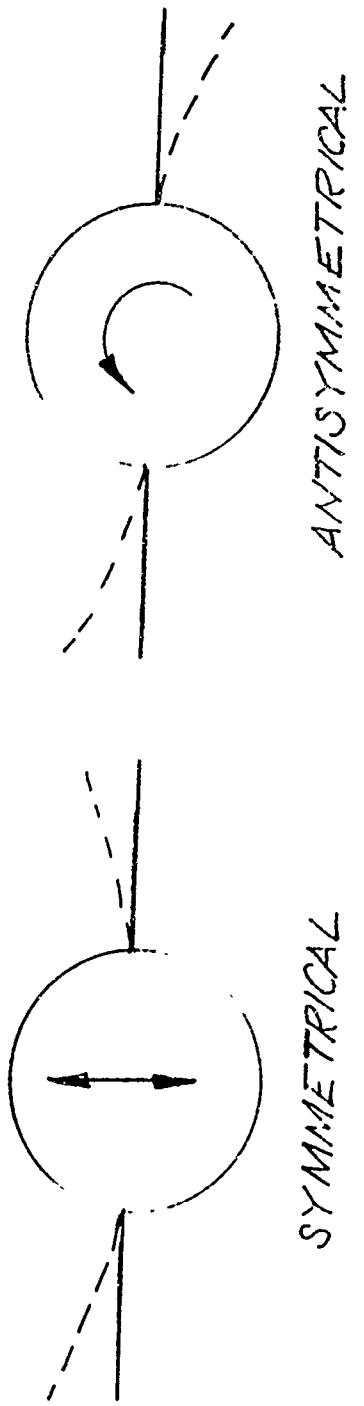


FIGURE A.2  
SAMPLE ELEMENT-TYPICAL MASSIVE  
FIXED WING-MINABLE CENTERLINE SURFACE

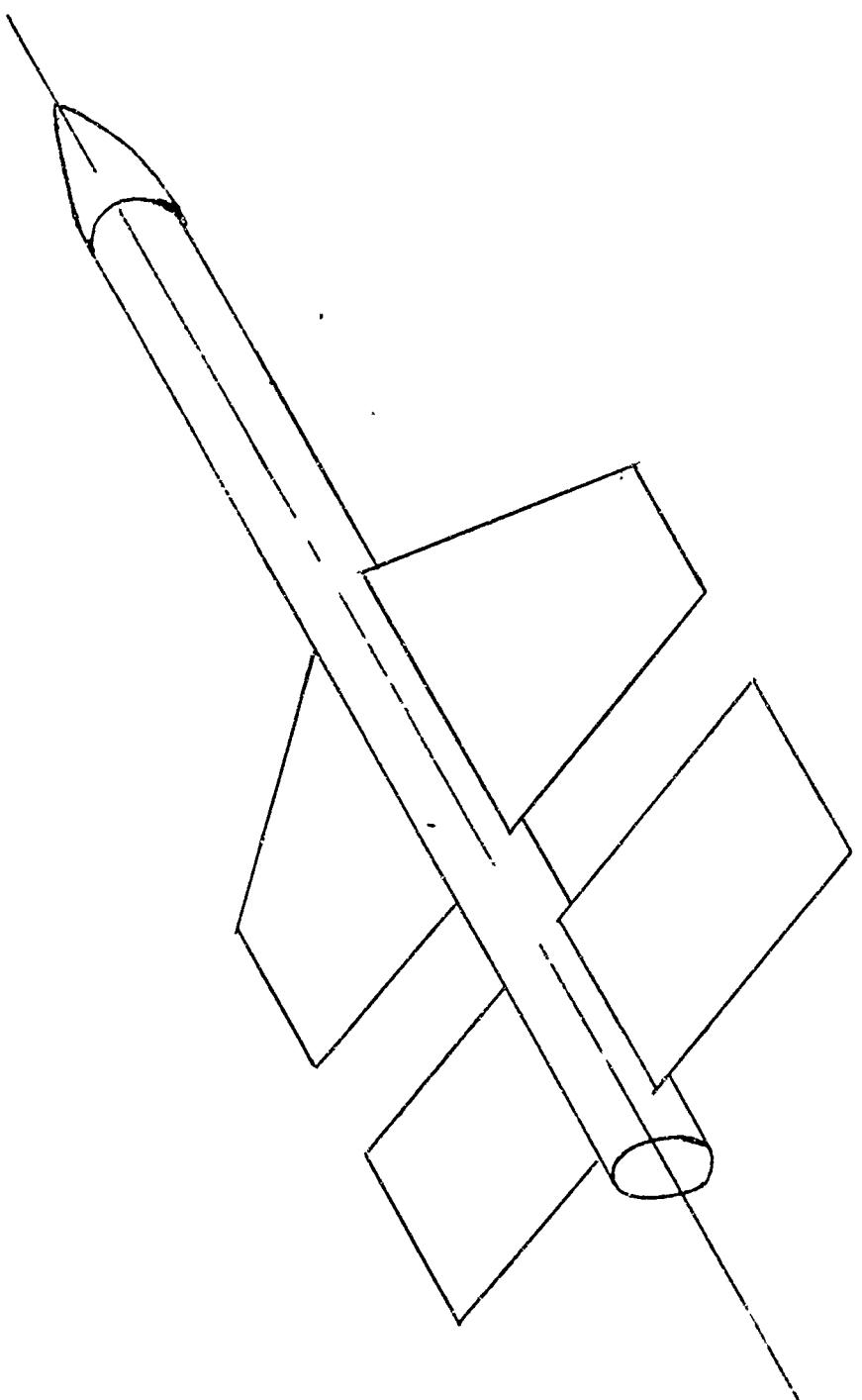
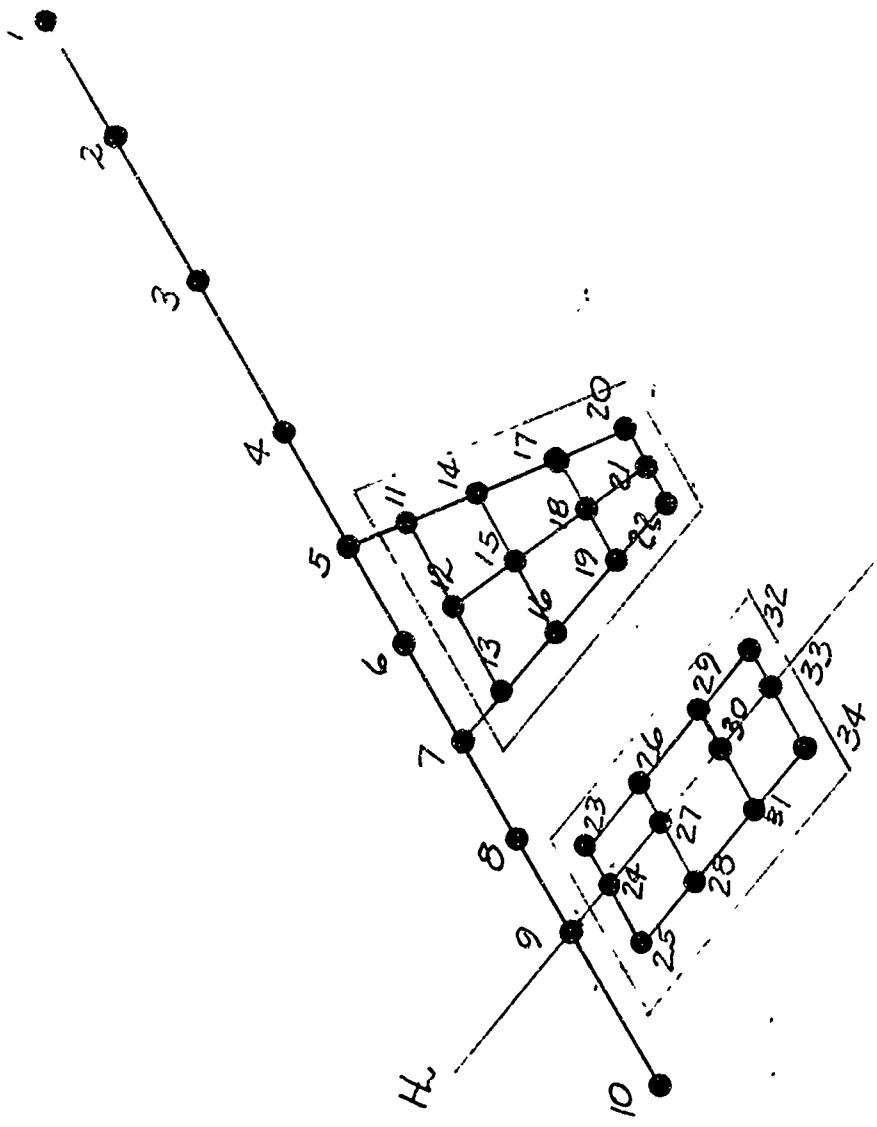


FIGURE 4.3  
MATHEMATICAL MODEL



## TYPICAL MISSILE WITH AERODYNAMICALLY COUPLED JANDEN SURFACES

NJTS = 34 NR = 10 NRE = 46 NPE = 0 NNODE = 5 HREY = 1 NLOMP = 34

MATERIAL PROPERTIES			
NO.	YOUNG'S MODULUS	POISSON'S RATIO	DENSITY
1	0.1000E 08	0.30000	0.36462E 07
2	0.4000E 08	0.30000	0.15385E 08

JOINT MATERIAL PROPERTIES			
JOINT NO.	X COORD.	Y COORD.	Z COORD.
1	0.	0.	0.
2	15.00000	0.	0.
3	25.00000	0.	0.

JOINT MATERIAL PROPERTIES			
JOINT NO.	X COORD.	Y COORD.	Z COORD.
4	35.00000	0.	0.
5	45.00000	0.	0.
6	55.00000	0.	0.
7	65.00000	0.	0.
8	75.00000	0.	0.
9	85.00000	0.	0.
10	100.00000	0.	0.
11	146.05000	5.00000	5.00000
12	155.62500	5.00000	5.00000
13	165.20800	5.00000	5.00000
14	174.13000	15.00000	15.00000
15	184.87500	154.00000	154.00000
16	193.62500	15.00000	15.00000
17	203.20500	25.00000	25.00000
18	212.12500	25.00000	25.00000
19	220.04900	25.00000	25.00000
20	227.29000	35.00000	35.00000
21	239.37500	35.00000	35.00000
22	266.45500	35.00000	35.00000
23	281.66000	5.00000	5.00000
24	285.00000	5.00000	5.00000
25	291.66000	5.00000	5.00000
26	291.66000	15.00000	15.00000
27	295.00000	15.00000	15.00000
28	291.66000	15.00000	15.00000
29	291.66000	25.00000	25.00000
30	295.00000	25.00000	25.00000
31	291.66000	25.00000	25.00000
32	291.66000	35.00000	35.00000
33	295.00000	35.00000	35.00000
34	291.66000	35.00000	35.00000

JOINT RESTRAINT COORDINATES			
JOINT NO.	Z DISPLACEMENT	Z ROTATION ABOUT X	Z ROTATION ABOUT Y
1	0	1	0
2	0	1	0
3	0	1	0
4	0	1	0

COORDINATE NUMBERS FOR EACH Z DISPLACEMENT AT EACH UNRESTRAINED JOINT

JOINT NO. COORD. NO.

1	1
2	1
3	1
4	1
5	1
6	1
7	1
8	1
9	1
10	1
11	1
12	1
13	1
14	1
15	1
16	1
17	1
18	1
19	1
20	1
21	1
22	1
23	1
24	1
25	1
26	1
27	1
28	1
29	1
30	1
31	1
32	1
33	1
34	1

LUMPE D. WEIGHT

JOINT NO.	WEIGHT
1	25.0000
2	25.0000
3	50.0000
4	50.0000
5	50.0000
6	50.0000
7	75.0000
8	75.0000
9	25.0000
10	25.0000
11	25.3330
12	25.3330
13	25.3330

1.4	0.2880
1.5	0.2880
1.6	0.2880
1.7	0.2550
1.8	0.2550
1.9	0.2550
2.0	0.2220
2.1	0.2220
2.2	0.2220
2.3	0.0840
2.4	0.1250
2.5	0.0410
2.6	0.0840
2.7	0.1250
2.8	0.0410
2.9	0.0840
3.0	0.1250
3.1	0.0410
3.2	0.0840
3.3	0.1250
3.4	0.0410

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B.E.A.M. ELEMENT PROPERTIES						
ELEMENT NO.	A	I	J	MAT	JOINT 1	JOINT 2
1	1.5000	35.0000	70.0000	1	1	2
2	1.5000	35.0000	70.0000	1	2	3
3	1.5000	35.0000	70.0000	1	3	4
4	3.0000	70.0000	140.0000	1	4	5
5	3.0000	70.0000	140.0000	1	5	6
6	3.0000	70.0000	140.0000	1	6	7
7	3.0000	70.0000	140.0000	1	7	8
8	3.0000	70.0000	140.0000	1	8	9
9	1.5000	35.0000	70.0000	1	9	10
10	0.5000	0.2660	0.0266	2	5	11
11	1.5000	0.2660	0.0266	2	11	14
12	0.5000	0.2660	0.0266	2	14	17
13	0.5000	0.2660	0.0266	2	17	20
14	0.5000	0.2660	0.0266	2	12	15
15	0.5000	0.2660	0.0266	2	15	18
16	0.5000	0.2660	0.0266	2	18	21
17	0.5000	0.2660	0.0266	2	7	13
18	0.5000	0.2660	0.0266	2	13	16
19	0.5000	0.2660	0.0266	2	16	19
20	0.5000	0.2660	0.0266	2	19	22
21	0.5000	0.2660	0.0266	2	11	12
22	0.5000	0.2660	0.0266	2	12	13
23	0.5000	0.2660	0.0266	2	14	15
24	0.5000	0.2560	0.0266	2	15	16
25	0.5000	0.2660	0.0266	2	17	18
26	0.5000	0.2660	0.0266	2	18	19
27	0.5000	0.2660	0.0266	2	20	21
28	0.5000	0.2660	0.0266	2	21	22
29	0.3000	0.8000	1.6000	1	9	24
30	0.5000	0.2660	0.0266	1	23	24
31	0.5000	0.2660	0.0266	1	24	25
32	1.5000	1.2660	0.0266	1	26	27
33	0.5000	0.2660	0.0266	1	27	28
34	0.5000	0.2660	0.0266	1	29	30
35	0.5000	0.2660	0.0266	1	30	31

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## REDUCED UPPER TRIANGULAR SLEEVNESS MATRIX

ROW 1	0.19748E 06	-0.57405E 06	0.48689E 06	-0.14534E 06	-0.12183E 05	-0.84334E 05	0.16937E 03	-0.11633E 02	-0.18876E 03
)	-0.34293E -01	-0.13173E -01	0.12946E -02	0.69176E -01	-0.21095E -00	-0.32066E -01	-0.38742E -01	-0.52917E -01	0.53236E -00
)	-0.25661E -02	0.94667E -02	0.45552E -00	-0.34060E 00	-0.11492E 00	0.48766E -03	-0.71414E -03	0.22936E -03	0.46933E -06
)	-0.51035E -05	0.47242E -05	0.64961E -06	-0.52872E -07	0.46376E -07				
ROW 2	0.22485E 07	-0.12601E 07	0.14534E 07	0.84183E 06	0.84034E 04	-0.16137E 04	0.11633E 03	0.16876E 04	-0.34993E 02
)	-0.13173E -02	-0.12946E -03	-0.69176E -00	-0.21095E -01	0.32066E -02	0.38742E -00	0.52917E -00	0.53236E -01	-0.25668E -01
)	-0.94667E -01	-0.45552E -01	0.34060E 01	0.11492E 01	-0.48476E -02	0.71414E -02	-0.22936E -02	-0.46933E -05	0.51035E -04
)	-0.47242E -04	-0.64961E -07	0.52872E -06	0.46376E -06					
ROW 3	0.51062E 07	-0.44405E 07	0.26864E 07	-0.54825E 06	-0.37815E 05	0.76216E 04	-0.52356E 03	-0.48942E 04	-0.15747E 03
)	-0.56253E 03	-0.31129E -01	0.94929E -01	-0.14439E -03	-0.17434E -01	-0.23068E -01	-0.24181E -02	0.11556E -00	-0.42688E -00
)	0.75102E 02	-0.15327E 02	-0.10689E 00	0.51714E 01	0.21814E -01	-0.32136E -01	0.10349E -03	0.114522E -02	-0.14324E -05
)	-0.41654E -04	-0.29232E -06	-0.23793E -05	0.20869E -05					
ROW 4	0.68485E 07	0.26864E 07	-0.18530E 06	-0.37346E 05	0.25651E 04	0.23981E 05	0.77159E 03	0.29947E 03	-0.28546E 04
)	-0.45253E 02	-0.46515E 02	0.70749E 03	0.85426E 01	0.11479E 02	-0.11849E 03	-0.566425E 00	-0.20974E 01	-0.18045E 01
)	0.75102E 02	-0.25340E 02	-0.10689E 00	0.15747E 00	-0.50578E -03	-0.10349E -03	0.114522E -02	-0.10417E -02	-0.14324E -05
)	-0.41654E -04	-0.29232E -06	-0.23793E -05	0.20869E -05					
ROW 5	0.90667E 07	0.30050E 07	-0.68565E 06	0.41600E 05	-0.28438E 05	-0.18683E 04	0.78533E 04	0.34884E 04	-0.36277E 01
)	-0.68377E -03	-0.84443E -03	-0.65325E -01	-0.17242E -01	0.14145E -03	-0.22824E -01	-0.28514E -02	-0.16289E -04	0.12108E -04
)	0.41095E 03	-0.17335E 01	0.25537E 01	-0.82025E 00	-0.16783E -02	0.16572E -01	-0.16889E -01	-0.23229E -01	0.18997E -03
)	-0.16824E -03								
ROW 6	0.83323E 07	-0.42514E 07	0.58049E 06	-0.21369E 04	0.66590E 03	-0.66539E 04	0.23232E 03	0.41649E 02	-0.13324E 01
)	-0.57983E 02	-0.33543E 02	-0.19524E 01	-0.16643E 03	-0.96878E 01	0.62335E 00	0.27234E 02	-0.22738E 05	0.16994E 02
)	-0.24189E 02	0.35635E 02	-0.11446E 02	-0.23419E 01	0.25915E 00	-0.23573E 00	-0.32415E 03	0.26363E 02	-0.23141E -02
)	0.38522E 05	0.62941E 03	-0.41621E 03	-0.94692E 04	-0.21211E 03	0.21206E 02	0.15756E 04	0.36128E 02	
ROW 7	0.34325E 07	-0.69944E 06	-0.43068E 03	-0.17455E 03	-0.13331E 04	-0.41649E 02	-0.41649E 01	-0.13324E 01	-0.11486E 02
)	0.79767E 06	0.33543E 02	-0.19524E 01	-0.16643E 03	-0.96878E 01	0.62335E 00	0.27234E 02	-0.22738E 05	0.16994E 02
)	-0.24189E 02	0.35635E 02	-0.11446E 02	-0.23419E 01	0.25915E 00	-0.23573E 00	-0.32415E 03	0.26363E 02	-0.23141E -02
)	0.38522E 05	0.62941E 03	-0.41621E 03	-0.94692E 04	-0.21211E 03	0.21206E 02	0.15756E 04	0.36128E 02	
ROW 8	0.24162E 06	-0.29582E 07	0.12264E 02	-0.91422E 02	0.32232E 01	0.26697E 00	0.91532E 01	0.80270E 00	-0.54769E 01
)	-0.23039E 01	0.13410E 00	-0.86294E 02	0.38405E 00	-0.27299E 04	-0.20544E 04	-0.68644E 03	0.28955E 01	-0.42657E 01
)	0.13701E 01	0.28034E 02	-0.31022E 01	0.24218E 01	0.36802E 04	-0.31502E 03	0.27761E 03		

ROW	9	0.71902E-06	-0.40051E-05	0.19036E-05	-0.20196E-06	0.84026E-03	-0.37711E-03	0.64956E-05	0.17683E-03	0.81226E-02	
)	-0.21067E-05	0.49346E-02	-0.30186E-02	0.11583E-01	-0.06610E-03	-0.29223E-00	0.12377E-02	-0.16168E-02	0.58328E-03	-0.11934E-02	
)	0.11934E-05	-0.13206E-04	0.12013E-04	0.16519E-07	-0.13445E-09	0.11793E-06					
)	0.96516E-05	-0.38363E-05	0.13363E-04	-0.11191E-05	0.14875E-04	0.17097E-03	0.25129E-05	-0.37443E-02	-0.22449E-02		
)	-0.42037E-04	0.17561E-02	-0.48021E-00	-0.35906E-00	0.12115E-00	-0.51104E-03	0.75285E-03	-0.24102E-03	-0.49477E-06		
)	0.54750E-05	-0.49803E-15	-0.68480E-08	0.55739E-07	-0.48890E-07						
)	ROW	11	0.76064E-06	-0.65086E-02	-0.33379E-03	-0.21396E-06	-0.53572E-02	0.69927E-03	0.68801E-05	0.18838E-02	-0.12305E-03
)	-0.11463E-05	0.35797E-01	-0.26766E-01	-0.90312E-00	0.38095E-02	-0.56121E-02	0.18026E-02	0.36838E-05	-0.40614E-04		
)	-0.37125E-04	0.51050E-07	-0.41550E-06	0.36445E-06							
)	ROW	12	0.16436E-06	-0.51049E-05	0.24255E-05	-0.10107E-06	0.22992E-04	-0.34927E-03	0.26597E-05	-0.43714E-03	0.75618E-02
)	-0.12621E-00	0.94370E-01	-0.1841E-01	0.31841E-01	-0.13331E-03	0.19707E-03	-0.63555E-04	-0.13004E-06	0.14390E-05	-0.13089E-05	
)	-0.17990E-06	0.14649E-07	-0.12849E-07								
)	ROW	13	0.20224E-06	-0.50733E-05	-0.16050E-04	-0.91492E-05	-0.20494E-04	0.25464E-02	0.25271E-05	-0.20385E-03	-0.11201E-01
)	-0.83754E-02	-0.28260E-02	-0.11920E-04	0.17561E-04	-0.56406E-05	-0.11541E-07	0.12771E-06	-0.11617E-06	-0.15974E-09		
)	-0.13002E-08	-0.11404E-08	-0.36489E-07								
)	ROW	14	0.19388E-06	-0.77602E-02	0.15963E-04	-0.10709E-06	-0.65459E-02	0.10508E-02	0.28277E-05	-0.35841E-00	0.26799E-00
)	-0.90421E-01	-0.38141E-03	-0.56182E-03	-0.18048E-03	-0.36928E-06	0.40863E-05	-0.32597E-06	0.44823E-09	-0.36482E-08	0.31999E-08	
)	-0.36489E-07										
)	ROW	15	0.13639E-06	-0.69122E-05	0.32645E-05	-0.35569E-05	0.34129E-04	-0.46723E-03	0.33431E-01	-0.23501E-01	-0.79295E-02
)	-0.33448E-04	-0.49275E-04	0.15827E-04	0.32384E-07	-0.35835E-06	0.32597E-06	0.44823E-09	-0.32597E-09	-0.36482E-08	0.31999E-08	
)											
)	ROW	16	0.23796E-06	-0.68975E-05	0.27417E-04	-0.31367E-05	0.29664E-04	0.21453E-02	-0.16041E-02	-0.54124E-03	0.22803E-05
)	-0.33634E-05	0.10803E-05	0.22104E-06	-0.24160E-07	0.22249E-07	0.30594E-10	-0.24901E-09	0.21842E-09			
)											
)	ROW	17	0.14224E-06	-0.10793E-03	-0.25542E-04	-0.41321E-05	-0.20214E-01	-0.67455E-01	-0.22760E-01	0.96005E-04	-0.14143E-03
)	-0.45428E-04	0.92950E-07	-0.10286E-05	0.93561E-06	-0.12865E-08	-0.10471E-07	0.91877E-08				
)											
)	ROW	18	0.64224E-05	-0.93278E-05	0.45224E-05	-0.52510E-02	0.32262E-02	0.13247E-02	-0.55880E-05	0.82322E-05	-0.26442E-05
)	-0.54101E-06	0.59867E-07	-0.54457E-07	-0.74883E-10	0.60948E-09	-0.53466E-09					
)											
)	ROW	19	0.20261E-06	-0.92801E-05	-0.33790E-03	-0.25265E-03	-0.85246E-04	-0.35958E-06	-0.52973E-06	-0.17015E-06	-0.34814E-02
)	-0.38524E-08	-0.35043E-08	-0.46187E-11	0.39220E-10	-0.34401E-10						
)											
)	ROW	20	0.64573E-05	-0.15038E-01	0.11244E-01	-0.37940E-02	-0.16004E-04	0.23572E-04	-0.75728E-05	-0.15494E-02	-0.17146E-02
)	-0.15596E-06	-0.21446E-09	0.17455E-08	-0.15311E-08							
)											
)	ROW	21	0.11438E-06	-0.13578E-06	0.26905E-05	-0.11891E-05	0.20158E-04	-0.16660E-04	0.48677E-02	-0.18985E-01	
)	-0.10753E-04	-0.84226E-01	0.33240E-00								
)											
)	ROW	22	0.53063E-06	-0.47280E-05	0.77597E-03	-0.67125E-05	0.32870E-02	0.52467E-03	0.19777E-05	0.26735E-03	-0.9076UE-02
)											

REDUCED UPPER TRIANGULAR FLEXIBILITY MATRIX									
ROW 1	-0.83073E-04	-0.43438E-04	-0.21776E-04	0.72572E-05	-0.16051E-05	-0.21194E-05	0.42198E-05	-0.241524E-06	
)	-0.29351E-06	-0.25922E-07	-0.11100E-05	-0.59161E-06	-0.41861E-07	-0.16806E-05	-0.92676E-06	-0.17898E-06	-0.21675E-05
)	-0.12731E-05	-0.35922E-06	-0.35644E-05	-0.42790E-05	-0.57038E-05	-0.35551E-05	-0.42797E-05	-0.57046E-05	-0.35661E-02
)	0.42807E-05	0.57056E-05	0.35567E-05	0.42818E-05	0.57066E-05				
)	ROW 2	0.2528E-04	0.13565E-04	0.42580E-05	-0.10701E-05	0.14263E-05	-0.28524E-05	-0.49920E-05	-0.19567E-06
)	0.13341E-07	-0.73998E-06	-0.3941E-06	-0.27908E-07	-0.11204E-05	-0.61784E-06	-0.11927E-06	-0.14583E-05	-0.84874E-06
)	-0.23951E-06	0.23763E-05	-0.20524E-05	-0.38026E-05	0.23767E-05	-0.28531E-05	-0.38031E-05	-0.23774E-05	-0.26238E-05
)	0.38037E-05	0.23761E-05	0.28545E-05	0.36044E-05					
)	ROW 3	0.80911E-05	0.30932E-05	0.21340E-06	0.95087E-06	0.19817E-05	0.33280E-05	-0.18455E-06	-0.13042E-06
)	-0.49332E-06	-0.26294E-06	-0.18605E-07	-0.74695E-06	-0.4190E-06	-0.79511E-07	-0.97233E-06	-0.56583E-06	-0.15968E-06
)	0.15842E-05	0.19018E-05	-0.25350E-05	0.15842E-05	0.19021E-05	0.25354E-05	0.15849E-05	-0.19125E-05	-0.25358E-05
)	0.15854E-05	0.19030E-05	0.25363E-05						
)	ROW 4	-0.14275E-05	-0.15670E-06	0.47544E-06	0.95088E-06	0.16640E-05	-0.92278E-07	-0.65224E-07	-0.44421E-08
)	-0.13147E-06	-0.93026E-08	-0.37348E-08	-0.20595E-06	-0.39756E-07	-0.18612E-06	-0.28292E-06	-0.79039E-07	-0.39209E-06
)	0.95009E-06	0.12675E-05	0.29225E-06	0.95105E-06	0.12677E-05	0.29247E-06	0.95128E-06	0.12679E-05	0.19272E-06
)	0.95151E-06	0.12682E-05							

ROW	5	0.23768E-06	-0.35573E-06	-0.71346E-06	-0.12486E-05	0.34716E-07	0.25249E-07	-0.54194E-08	0.92012E-07	0.44197E-07
	6	0.14277E-05	0.30935E-05	0.55921E-05	-0.146584E-07	-0.35771E-07	0.17230E-07	-0.1230E-06	-0.45315E-07	0.37667E-07
	7	-0.17610E-05	-0.63252E-05	0.48243E-07	-0.22071E-06	-0.83882E-07	0.52800E-07	0.23371E-05	0.30935E-05	0.42029E-05
	8	0.76152E-05	0.14755E-04	-0.93169E-07	-0.71592E-07	-0.34461E-07	-0.24277E-06	-0.90639E-07	0.75333E-07	-0.35222E-06
	9	0.16885E-06	-0.27225E-06	-0.44143E-06	-0.16776E-06	0.10566E-06	0.60257E-05	0.61566E-05	0.10786E-04	0.60270E-05
	10	0.36679E-04	0.21342E-05	-0.12520E-06	0.60306E-07	-0.42484E-06	-0.15868E-06	0.13163E-06	-0.61636E-06	-0.22140E-06
	11	0.36916E-05	0.28484E-04	-0.29358E-06	-0.18488E-06	-0.11258E-04	0.14756E-04	0.21731E-04	0.11269E-04	0.1458E-04
	12	0.36772E-05	0.21692E-05	0.79186E-05	0.94396E-05	0.73152E-05	0.10977E-04	0.2085E-04	0.14397E-04	0.52837E-05
	13	0.62750E-04	0.53788E-04	0.24277E-06	-0.32362E-04	-0.59596E-07	0.21543E-07	-0.93170E-07	-0.7762E-07	-0.12421E-06
	14	0.92695E-04	0.56012E-04	0.17312E-04	0.17368E-05	0.53575E-05	0.13941E-04	0.22528E-04	0.22553E-05	0.19989E-04
	15	0.10042E-06	0.622767E-07	0.75348E-07	0.10043E-06	-0.24261E-06	0.28712E-07	0.34467E-07	0.45943E-07	0.26720E-07
	16	0.23084E-03	0.23023E-05	0.37355E-05	0.57876E-05	0.57876E-05	0.37131E-03	0.16460E-03	0.2339E-06	0.46950E-06
	17	0.36669E-03	0.14575E-03	0.16508E-03	0.56344E-06	-0.12956E-05	-0.12956E-05	-0.35244E-06	-0.46973E-06	-0.10541E-05

55	ROW 18	0.12864E-05	1.80415E-05	0.96570E-07	0.12866E-06	0.80440E-07	0.96554E-07	0.12868E-06
56	ROW 19	0.9567E-03	0.61266E-03	0.27337E-03	0.36771E-06	0.3143E-06	0.52843E-06	0.36779E-06
57	ROW 20	-0.36789E-06	-0.44161E-06	-0.58461E-06	-0.37800E-06	-0.44172E-06	-0.58871E-06	-0.58851E-06
58	ROW 21	0.93067E-03	0.59980E-03	-0.13575E-06	-0.16771E-06	-0.22365E-06	-0.13374E-06	-0.22374E-06
59	ROW 22	0.14081E-06	0.88039E-07	0.10560E-06	0.14077E-06	0.87905E-07	0.10562E-06	0.14079E-06
60	ROW 23	0.23744E-04	0.11235E-04	-0.46898E-05	0.36326E-04	0.26560E-04	0.75275E-05	0.49936E-04
61	ROW 24	0.12825E-04	0.15995E-04	0.26888E-04	0.28457E-04	0.31628E-04	0.42504E-04	0.44094E-04
62	ROW 25	0.34403E-03	0.21467E-03	-0.21967E-04	0.95105E-04	0.65334E-05	0.37983E-04	0.14282E-04
63	ROW 26	0.73908E-03	0.54322E-04	0.46788E-03	0.12910E-02	0.18843E-03	0.71837E-03	0.17773E-02
64	ROW 27	0.13290E-02	0.95532E-03	-0.21259E-03	0.20101E-02	0.14889E-02	0.44760E-03	
65	ROW 28	0.96797E-03	0.97834E-03	0.15160E-02	0.15146E-02	0.15119E-02		
66	ROW 29	0.25054E-02	0.52507E-03	-0.45593E-02	0.36270E-02			
67	ROW 30	0.32860E-02	0.25098E-02	0.96514E-03				
68	ROW 31	0.25097E-02	0.24980E-02					
69	ROW 32	0.55671E-02						

UPPER TRIANGLE OF EXPANDED FLEXIBILITY MATRIX WITH ATTACH POINTS INCLUDED

ROW 1	0.83073E-04	0.43438E-04	0.21776E-04	0.72572E-05	0.	-0.16051E-05	0.	0.-21394E-05
	0.42789E-05	0.74980E-05	-0.41524E-06	-0.29351E-06	0.20012E-07	-0.11100E-05	-0.59161E-06	-0.-41861E-05
	-0.16806E-05	-0.92527E-06	-0.17890E-06	-0.21875E-05	-0.12731E-05	-0.35927E-06	-0.35643E-05	-0.-32790E-05
	0.57038E-05	0.35621E-05	0.42797E-05	0.57046E-05	0.35661E-05	0.42867E-05	0.57056E-05	0.-35672E-05
ROW 2	0.42818E-05	0.57206E-05						
	0.25228E-04	0.13565E-04	0.47648E-05	0.	0.10201E-05	0.	0.-14263E-05	0.-28520E-05
	0.49920E-05	-0.7683E-06	-0.19567E-06	0.13341E-07	-0.73998E-06	-0.39441E-06	-0.27908E-07	-0.-11204E-05
	-0.61784E-06	-0.11922E-06	-0.14583E-05	-0.84874E-06	-0.23951E-06	0.23763E-05	0.26526E-05	0.-38026E-05
	0.23767E-05	0.28531E-05	0.38031E-05	0.23774E-05	0.28538E-05	0.38037E-05	0.23781E-05	0.-28545E-05
	0.38044E-05							
ROW 3	0.80911E-05	-0.30932E-05	0.	-0.71340E-06	0.	0.25087E-06	0.19017E-05	0.-33280E-05
	-0.18455E-06	-0.3045E-06	0.86942E-08	-0.49332E-06	-0.26294E-06	-0.18605E-07	-0.74695E-06	-0.-41194E-05
	-0.79511E-07	-0.92723E-06	-0.56503E-16	-0.15968E-06	0.15842E-05	0.19018E-05	0.23350E-05	0.-15845E-05
	-0.19021E-05	0.25354E-05	0.15849E-05	0.19025E-05	0.25358E-05	0.19545E-05	0.19350E-05	0.-25363E-05
ROW 4	0.14275E-05	0.	-0.35670E-06	0.	0.47544E-06	0.95080E-06	0.16840E-05	-0.-2278E-05
	-0.65224E-07	0.44471E-08	-0.24666E-06	-0.13147E-06	-0.93026E-08	-0.37348E-06	-0.2095E-06	-0.-39156E-07
	-0.48612E-06	-0.28222E-06	-0.79839E-07	0.79209E-06	0.95089E-06	0.12675E-05	0.79225E-06	0.-29105E-06
	0.12677E-05	0.79242E-06	0.95128E-06	0.12679E-05	0.79272E-06	0.95151E-06	0.12682E-05	
ROW 5	0.	0.	0.	0.	0.	0.	0.	
	0.	0.	0.	0.	0.	0.	0.	
	0.	0.	0.	0.	0.	0.	0.	
	0.	0.	0.	0.	0.	0.	0.	
ROW 6	0.	0.	0.	0.	0.	0.	0.	
	0.23788E-06	0.	-0.35673E-06	-0.71346E-06	-0.12486E-05	0.34716E-07	0.25249E-07	-0.-54194E-08
	-0.92012E-07	0.44197E-07	-0.70910E-08	0.13740E-06	0.62302E-07	-0.21219E-08	0.17671E-06	-0.-91709E-07
	0.67599E-08	-0.59433E-06	-0.71348E-06	-0.95106E-06	-0.59415E-06	-0.71360E-06	-0.95119E-06	-0.-59461E-06
	-0.71376E-06	-0.29535E-06	-0.59479E-06	-0.71394E-06	-0.95152E-06			
ROW 7	0.	0.	0.	0.	0.	0.	0.	
	0.	0.	0.	0.	0.	0.	0.	
	0.	0.	0.	0.	0.	0.	0.	
	0.	0.	0.	0.	0.	0.	0.	
ROW 8	0.	0.	0.	0.	0.	0.	0.	
	0.14277E-05	0.36955E-05	0.54021E-05	-0.46584E-07	-0.35771E-07	0.17230E-07	-0.-12130E-07	0.-45112E-07
	0.37667E-07	-0.17610E-06	-0.63252E-07	0.-48243E-07	-0.22071E-06	-0.83802E-07	0.52800E-07	0.-25371E-07
	0.30935E-05	0.-20229E-05	0.26376E-05	0.30940E-05	0.42035E-05	0.25384E-05	0.30948E-05	0.-42042E-05
	0.25399E-05	0.-10952E-05	0.421149E-05					

ROW 9	$0.76155E-05$	$0.14756E-04$	$-0.93169E-07$	$-0.71542E-07$	$0.34616E-07$	$-0.24277E-06$	$-0.90630E-07$	$0.75553E-07$
	$-0.35221E-06$	$-0.12651E-06$	$-0.96487E-07$	$-0.44143E-06$	$-0.16776E-06$	$-0.10569E-06$	$0.60257E-05$	$0.76155E-05$
	$-0.10786E-04$	$0.60270E-05$	$0.76169E-05$	$0.10787E-04$	$0.60287E-05$	$0.76167E-05$	$0.10789E-04$	$0.60307E-05$
	$0.76206E-05$	$-0.10791E-04$						
ROW 10								
	$-0.23567E-04$	$-0.16305E-06$	$-0.12520E-06$	$-0.60306E-07$	$-0.42484E-06$	$-0.15860E-06$	$-0.13183E-06$	$-0.61636E-06$
	$-0.22148E-06$	$0.16886E-06$	$-0.77250E-06$	$-0.29358E-06$	$0.88480E-06$	$0.11258E-04$	$0.14756E-04$	$0.21731E-04$
	$-0.11268E-04$	$0.14758E-04$	$-0.21734E-04$	$0.11264E-04$	$0.11262E-04$	$0.21737E-04$	$0.11267E-04$	$0.14766E-04$
	$0.21741E-04$							
ROW 11								
	$0.38916E-05$	$0.21342E-05$	$-0.26835E-06$	$0.14387E-04$	$8.822794E-05$	$0.21478E-05$	$0.23509E-04$	$0.14397E-04$
	$-0.52237E-05$	$0.31876E-04$	$0.20464E-04$	$0.90874E-05$	$-8.77611E-07$	$-0.93170E-07$	$-0.12420E-06$	$-0.77626E-07$
	$-0.93106E-07$	$-0.12421E-06$	$-0.77648E-07$	$-0.93208E-07$	$-0.12423E-06$	$-0.77672E-07$	$-0.93231E-07$	$-0.12426E-06$
ROW 12								
	$0.13711E-04$	$8.19517E-05$	$-0.79186E-05$	$0.94396E-05$	$9.7352E-05$	$0.16977E-04$	$0.10205E-04$	$0.10251E-04$
	$-0.13114E-04$	$0.12684E-04$	$0.12362E-04$	$0.59296E-07$	$-9.7533E-07$	$-0.93366E-07$	$-0.59607E-07$	$-0.71555E-07$
	$-0.95368E-07$	$-0.59624E-07$	$-0.71572E-07$	$-0.95396E-07$	$-0.59642E-07$	$-0.71599E-07$	$-0.95413E-07$	
ROW 13								
	$0.36776E-05$	$0.21696E-05$	$0.79270E-05$	$0.13698E-04$	$6.53575E-05$	$0.13941E-04$	$6.22528E-04$	$6.92553E-05$
	$-0.19989E-04$	$-0.30727E-04$	$-0.28746E-07$	$-0.34461E-07$	$-0.45936E-07$	$-0.28712E-07$	$-0.34467E-07$	$-0.45943E-07$
	$-0.26724E-07$	$0.34475E-07$	$0.45951E-07$	$0.28792E-07$	$0.34494E-07$	$0.45959E-07$		
ROW 14								
	$0.92995E-04$	$0.56012E-04$	$8.17312E-04$	$-0.17368E-03$	$-0.10746E-03$	$0.41823E-04$	$0.24744E-03$	$0.15962E-03$
	$-0.21792E-04$	$-0.20223E-06$	$-0.24227E-06$	$-0.32361E-06$	$-0.20227E-06$	$-0.24281E-06$	$-0.32366E-06$	$-0.20233E-06$
	$-0.24287E-06$	$-0.32371E-06$	$-0.20239E-06$	$-0.24293E-06$	$-0.32377E-06$			
ROW 15								
	$0.62750E-04$	$0.53786E-04$	$8.11335E-03$	$0.11295E-03$	$8.16895E-03$	$0.16660E-03$	$0.16365E-03$	$0.62754E-03$
	$-0.2596E-07$	$-0.90631E-07$	$-0.12041E-06$	$-0.75511E-07$	$-0.92634E-07$	$-0.12083E-06$	$-0.75531E-07$	$-0.20233E-06$
	$-0.12005E-06$	$-0.75534E-07$	$-0.90689E-07$	$-0.12087E-06$				
ROW 16								
	$0.38859E-04$	$0.42017E-04$	$8.10416E-03$	$0.16699E-03$	$8.72556E-04$	$0.15582E-03$	$0.23919E-03$	$0.62754E-03$
	$0.75335E-07$	$-0.10442E-06$	$-0.62267E-07$	$-0.25348E-07$	$-0.19843E-06$	$0.42278E-02$	$0.75336E-07$	$0.19945E-06$
	$-0.62884E-07$	$0.75305E-07$	$0.10847E-06$					
ROW 17								
	$0.37984E-03$	$0.23767E-03$	$8.97406E-04$	$0.57876E-03$	$0.37131E-03$	$0.16468E-03$	$0.29339E-04$	$0.62754E-03$
	$-0.46939E-06$	$-0.29345E-06$	$-0.35227E-06$	$-0.46955E-06$	$-0.29351E-06$	$-0.46964E-06$	$-0.29366E-06$	$-0.29367E-06$
	$-0.10541E-06$	$-0.17654E-06$	$-0.16867E-06$					
ROW 18								
	$0.22684E-03$	$0.23093E-03$	$8.37355E-03$	$0.36895E-03$	$8.36380E-03$	$0.10539E-03$	$0.12692E-03$	$0.16864E-03$
	$-0.10541E-06$	$-0.17654E-06$	$-0.16867E-06$					

-0.16872E-06.

ROW 19

0.36669E-03	0.16575E-03	0.36308E-03	0.56147E-03	0.63375E-03	0.96488E-03	0.12842E-03	0.85992E-03	0.96354E-03	0.12898E-03
0.96515E-07	0.12864E-06	0.6n415E-07	0.96529E-07	0.12H66E-06	0.12H66E-06	0.804440E-07	0.96354E-07	0.12898E-06	0.12898E-06

ROW 20

0.95567E-03	0.61266E-03	0.27337E-03	0.36771E-03	0.44115E-03	0.58043E-03	0.58043E-03	0.73618E-03	0.44152E-03	0.44152E-03
-0.588851E-06	-0.36709E-06	-0.44161E-06	-0.58861E-06	-0.73618E-06	-0.44172E-06	-0.58871E-06	-0.73621E-06	-0.44172E-06	-0.58871E-06

ROW 21

0.646942E-03	0.59980E-03	0.11975E-03	0.16777E-03	0.21163E-03	0.21163E-03	0.16787E-03	0.21163E-03	0.16787E-03	0.21163E-03
-0.13981E-03	-0.16783E-03	-0.16783E-03	-0.27370E-03	-0.13985E-03	-0.13985E-03	-0.16787E-03	-0.22374E-03	-0.16787E-03	-0.22374E-03

ROW 22

0.93067E-03	0.87966E-03	0.16560E-03	0.14077E-03	0.87655E-03	0.10962E-03	0.14077E-03	0.87655E-03	0.10962E-03	0.14077E-03
-0.10545E-03	-0.14081E-03	-0.14081E-03	-0.87619E-03	-0.10967E-03	-0.10967E-03	-0.14081E-03	-0.22374E-03	-0.14081E-03	-0.22374E-03

ROW 23

0.23744E-04	0.11235E-04	-0.46090E-05	0.36326E-04	0.24369E-04	0.24369E-04	0.36326E-04	0.24369E-04	0.36326E-04	0.24369E-04
0.20469E-04	0.63575E-04	0.53546E-04	0.33562E-04	-	-	-	-	-	-

ROW 24

0.12025E-04	0.15995E-04	0.26668E-04	0.28457E-04	0.31628E-04	0.42594E-04	0.42594E-04	0.44694E-04	0.44694E-04	0.44694E-04
0.56141E-04	0.29731E-04	0.64996E-04	0.64996E-04	-	-	-	-	-	-

ROW 25

0.92993E-04	-0.10219E-05	0.3n420E-04	0.95105E-04	0.45134E-03	0.37984E-03	0.10871E-03	0.42022E-03	0.67494E-03	0.67494E-03
0.44703E-04	0.10566E-03	-	-	-	-	-	-	-	-

ROW 26

0.34483E-03	0.21467E-03	-0.21967E-04	0.62871E-03	0.47106E-03	0.1n571E-03	0.47106E-03	0.68894E-03	0.68894E-03	0.47106E-03
0.97524E-04	-	-	-	-	-	-	-	-	-

ROW 27

0.21462E-01	-0.22524E-01	0.47269E-01	0.41012E-01						
-	-	-	-	-	-	-	-	-	-

ROW 28

0.73996E-03	0.74322E-04	0.44788E-03	0.12910E-02	0.10643E-02	0.10643E-02	0.12910E-02	0.10643E-02	0.12910E-02	0.10643E-02
-	-	-	-	-	-	-	-	-	-

ROW 29

0.13298E-02	0.95377E-03	0.21250E-03	0.20101E-02	0.14069E-02	0.14069E-02	0.20101E-02	0.14069E-02	0.20101E-02	0.14069E-02
-	-	-	-	-	-	-	-	-	-

ROW 30

0.95272E-01	0.78144E-01	0.15115E-02	0.25144E-02	0.15115E-02	0.25144E-02	0.15115E-02	0.25144E-02	0.15115E-02	0.25144E-02
-	-	-	-	-	-	-	-	-	-

ROW 31

0.25024E-03	50 E-01	0.176274E-02	7.36274E-02	-	-	-	-	-	-
-	-	-	-	-	-	-	-	-	-

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 ) ROW 32 0.32660E-02 0.75098E-02 0.96514E-03  
 ) ROW 33

 ) 0.25097E-02 0.24980E-02  
 ) ROW 34

0.55671E-02

## REDUCED UPPER TRIANGULAR WEIGHT MATRIX

ROW 1	0.25000E-02	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
ROW 2	0.25000E-02	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
ROW 3	0.50000E-02	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
ROW 4	0.50000E-02	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
ROW 5	0.50000E-02	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
ROW 6	0.75000E-02	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
ROW 7	0.75000E-02	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.
ROW 8	0.25000E-02	0.	0.	0.	0.	0.	0.
) 0.	0.	0.	0.	0.	0.	0.	0.



886872

HERE ARE THE EIGENVALUES AND EIGENVECTORS.

EIGENVECTOR NUMBER 1

CORRESPONDING TO 1.2431656E 05

1.000000E 00	5.9942504E 01	2.895606E 01	1.0198256E 01	2.5693890E 02	4.7414512E 02
1.049420E 01	1.9933320E 01	-6.3802740E 03	-4.5111410E 03	3.5171355E 04	-1.813797E 02
-9.3251658E 03	-2.3621349E 02	-1.6221170E 02	-3.9874489E 03	-3.8304938E 02	
-2.3506465E 02	-7.7221027E 03	8.571401E 02	1.058023E 01	1.4553430E 01	9.8412937E 02
1.1959757E 01	-1.6289247E 01	-1.1684120E 01	1.3910497E 01	1.8331725E 01	1.3777984E 01
1.6027544E 01	2.0511256E 01				

EIGENVECTOR NUMBER 2

CORRESPONDING TO 2.8039211E 05

-1.5489999E 01	-6.4672757F 02	-1.8471959E 02	3.9698923E 02	-2.7664241E 02	9.5851244E 02
2.1696881E 01	5.2824663E 01	-1.6082862E 03	-1.2945356E 03	-1.0799532E 03	-4.5213562E 03
-1.081705E 03	2.7339503E 03	-6.8087791E 03	-1.356056E 03	4.0358022E 03	-6.7062718E 03
-1.8082248E 03	5.0825463E 03	-2.0374804E 01	2.6093804E 01	3.6981248E 01	3.4374429E 01
4.0913597E 01	5.5092866E 01	5.4776460E 01	6.216722E 01	7.6713595E 01	7.8061574E 01
8.53099114E 01	1.0000000E 00				

EIGENVECTOR NUMBER 3

CORRESPONDING TO 4.3326807E 05

-2.1796538E 03	6.5790983E 04	-5.32E 05	-2.80292817E 04	4.7622906E 04	-2.4253131E 03
-6.3457642E 02	-1.4308372E 02	5.1749603E 05	3.9357314E 05	-2.8699178E 05	1.4919521E 04
-4.06666132E 05	-7.7678451E 05	2.3026971E 04	5.46660994E 05	-1.1925866E 04	2.9969769E 04
7.3603010E 05	-1.5379343E 04	1.5759983E 02	1.5922906E 02	6.1527410E 03	2.5555960E 01
2.5235382E 01	2.7953943E 01	6.0323156E 01	6.0790122E 01	6.1417854E 01	1.0000000E 00
9.9356139E 01	9.8001037E 01				

EIGENVECTOR NUMBER 4

CORRESPONDING TO 6.2475530E 05

9.7551997E 04	1.6620259E 04	-1.7080596E 04	-2.3098697E 04	8.9622011E 05	2.2531093E 05
1.0694025E 04	3.0328284E 04	3.6025923E 02	2.4747506E 02	3.4522297E 02	2.7336800E 01
2.8251094E 01	2.6233973E 01	6.193248E 01	6.0975550E 01	5.9592726E 01	1.0000000E 00
9.8168848E 01	9.6441956E 01	7.0102512E 05	1.0493107E 04	1.79521359E 04	3.8247457E 05
8.1975391E 05	1.8664724E 04	-1.7349771E 06	4.8575347E 05	1.4940868E 04	-4.6776962E 05
1.0983098E 05	1.2598648E 04				

EIGENVECTOR NUMBER 5

CORRESPONDING TO 1.2676136E 06

1.3755801E 04	2.3155412E 05	-7.7436289E 05	-6.2546180E 05	5.7484782E 05	-2.2257886E 04
-3.9333757E 04	-6.6603086E 04	9.2045341E 06	5.0705673E 06	-6.5691127E 06	3.0424150E 05
-8.4378481E 05	-3.4136023E 05	5.9234344E 05	-5.8720263E 06	-7.9847655E 05	8.3280020E 05
-1.2677529E 05	-1.087199AE 04	-9.7864893E 03	-4.7456958E 04	2.3681893E 02	-1.8488673E 01
-1.025687E 03	3.6595235E 01	-3.4812518E 01	1.5045557E 04	6.2936927E 01	-4.9240754E 01
5.3561601E 03	1.0000000E 00				

HERE ARE THE NATURAL FREQUENCIES

THE NATURAL FREQUENCY NUMBER	1	IS	56.116	CPS
THE NATURAL FREQUENCY NUMBER	2	IS	84.276	CPS
THE NATURAL FREQUENCY NUMBER	3	IS	104.763	CPS
THE NATURAL FREQUENCY NUMBER	4	IS	125.796	CPS
THE NATURAL FREQUENCY NUMBER	5	IS	179.196	CPS

## HUGHES AIRCRAFT CO. SUBSONIC AIC PROGRAM

## FLIGHT CONDITIONS AND GEOMETRY

MACH NUMBER = 0.50000 SPEED OF SOUND = 13480.000 L/T

RHO=0.10000000E 01

	WING	TAIL
L.E. STATION (L)	0.	40.000
ROOT CHORD (L)	30.000	15.000
L.E. SPAN (L)	0.	35.000
T.E. SPAN (L)	35.000	35.000
TIP CHORD (L)	20.000	15.000
TOTAL AREA (L*L)	1750.000	1050.000
SPAN COLL. STA.	4	4
CHORD COLL. STA.	3	3
CHORD INTG. STA.	18	18
SPAN PRES MODES	4	4
CHORD PRES MODES	3	3

## HUGHES AIRCRAFT CO. SUPERSONIC AIC PROGRAM (CONT-D)

OSCILLATORY FREQUENCY (CFS) 7.15136E-01

REFERENCE CHORD 1.50000E-01

REDUCED FREQUENCY (REF. CHORD) 1.00000E-00

REDUCED VELOCITY (REF. CHORD) 1.00000E-00

FREE STREAM MACH NUMBER 5.00000E-01

FREE STREAM VELOCITY 6.74000E-03

DENSITY 1.00

DYNAMIC PRESSURE (1/2\*RH0\*VEL\*\*2) 2.27130E-07

## AERODYNAMIC INFLUENCE COEFFICIENTS

RL IN RL

$ROW = 1$

2.6152E-00	-7.3687E-01	-4.8823E-06	-3.2104E-01	3.0947E-00	3.1773E-02	-6.0182E-00	1.6042E-00	1.1440E-01	-3.3341E-01
-5.4858E-00	-5.7732E-02	4.2262E-00	-9.7706E-01	-6.7701E-01	3.0536E-01	3.0421E-00	-2.3376E-01	-1.0421E-01	1.9187E-01
-1.6664E-00	1.0922E-03	-7.1213E-01	-8.6658E-04	-2.9557E-01	9.0611E-02	6.0997E-01	-1.5677E-01	-3.2660E-01	-2.1251E-03
-3.8379E-02	3.2667E-02	-6.2473E-02	-4.3454E-02	-2.3371E-02	-1.5552E-01	7.7996E-01	-2.9810E-01	-1.4365E-01	
-1.6693E-01	3.3865E-02	-1.1589E-02	9.1471E-03	2.4096E-02	-1.7101E-02	-1.4160E-02	5.3565E-03		

$ROW = 2$

-1.0936E-00	-2.7342E-03	3.9008E-00	3.0083E-00	-1.8168E-00	-3.4271E-00	2.6669E-00	6.2186E-02	-5.7566E-00	-3.6167E-00
2.1025E-00	3.6153E-00	-1.9480E-00	-6.8767E-02	4.5326E-00	2.3169E-00	-1.9988E-00	-2.4544E-00	4.4641E-01	3.7177E-02
-8.9782E-01	-5.2946E-01	3.4812E-01	4.8335E-01	-7.6720E-02	2.5152E-01	1.7725E-01	-4.9636E-01	-1.5122E-01	2.2487E-01
-1.1350E-02	5.9329E-02	2.6955E-02	-1.1627E-01	-2.7331E-02	5.4249E-02	-3.5491E-02	1.6357E-01	8.4514E-02	-3.2211E-01
-8.1975E-02	1.4924E-01	-1.6009E-03	1.5618E-02	4.5075E-03	-3.0837E-03	-1.0761E-02	-6.0436E-03	1.4706E-02	

$ROW = 3$

-1.8910E-00	6.5292E-01	5.68224E-00	1.0458E-00	-3.5902E-00	-1.8191E-00	4.9241E-00	-1.4114E-00	-1.0195E-01	-7.3014E-01
-1.9910E-00	-1.9440E-00	-3.2432E-00	8.5815E-01	6.8259E-00	3.0930E-01	-3.4566E-00	-1.1282E-00	7.9806E-01	-1.4598E-01
-1.5538E-00	-1.0840E-01	7.2164E-01	2.8566E-01	1.9261E-01	1.1684E-01	-3.0860E-01	-2.4860E-01	1.6209E-01	1.7365E-01
-2.9277E-02	2.1422E-02	-5.3739E-02	-4.4178E-02	2.0385E-02	2.7375E-02	1.0430E-01	6.4680E-02	-2.0304E-01	1.3655E-01
-8.4518E-02	9.3000E-02	9.5018E-03	5.0122E-03	-1.8626E-02	-1.0761E-02	7.9715E-03	7.7022E-03		

$ROW = 4$

1.3045E-00	-3.7374E-01	-2.3091E-00	-2.2811E-01	1.3927E-00	1.2970E-02	-3.3846E-00	7.7892E-01	5.7614E-00	-1.6192E-01
-2.8097E-00	-6.2022E-02	2.9544E-00	-6.8535E-01	-4.6499E-00	1.7269E-01	2.0175E-01	-1.3141E-01	-7.2923E-01	1.3192E-01
-1.5538E-00	7.5244E-03	-4.9881E-01	-5.9817E-01	-1.5964E-01	1.0047E-01	3.3272E-01	-1.8739E-01	-1.9152E-01	4.9865E-02
-3.0091E-02	2.9828E-02	6.1949E-02	-5.7347E-02	-3.7872E-02	2.1099E-02	-9.3715E-02	7.5502E-02	1.9618E-01	-1.4292E-01
-1.1675E-01	4.5701E-02	-8.513E-03	6.1606E-03	1.8202E-02	-1.5459E-02	-1.1130E-02	5.2387E-03		

$ROW = 5$

-5.5837E-01	-6.5447E-03	2.1154E-00	1.5454E-00	-1.0370E-00	-1.8195E-00	1.2668E-00	1.4596E-02	-2.7109E-00	-1.6918E-00
9.7790E-01	1.6849E-00	-1.4241E-00	-5.6690E-02	3.3526E-00	1.6952E-00	-1.5040E-00	-1.8093E-00	3.0567E-01	2.5222E-02
-6.02633E-01	-3.5473E-01	2.3030E-01	3.1900E-01	-7.0944E-02	1.9668E-01	1.5830E-01	-1.2735E-01	1.7244E-01	

-1.5893E-02 4.8666E-02 3.5788E-02 -9.5365E-02 -2.9622E-02 4.2887E-02 -4.2658E-02 1.3192E-01 9.6479E-02 -2.5902E-01  
 -8.0377E-02 1.1651E-01 -3.3315E-03 1.2934E-02 7.7723E-03 \*2.5441E-02 -7.0422E-03 1.1642E-02  
  
 ROW = 6  
 -9.5431E-01 5.2669E-01 2.9214E 00 5.3542E-01 -1.8581E 00 -9.5127E-01 2.4058E 00 -6.9928E-01 -4.9689E 00 -3.1162E-01  
 2.4447E 00 9.0898E-01 -2.3041E 00 5.9203E-01 4.8823E 00 2.8517E-01 -2.4693E 00 -8.6490E-01 5.5620E-01 -9.7584E-02  
 -1.0688E 00 -1.3674E-01 4.9171E-01 1.9544E-01 1.0471E-01 8.8257E-02 -2.0473E-01 -1.8413E-01 6.0392E-02 1.1803E-01  
 -2.0992E-02 1.9068E-02 -3.9699E-02 -3.9268E-02 1.4782E-02 2.4110E-02 6.5099E-02 5.4146E-02 -1.2666E-01 -1.1265E-01  
 4.9747E-02 7.2010E-02 6.5060E-03 4.7509E-03 -1.2782E-02 -9.9924E-03 5.2025E-03 6.6257E-03  
  
 ROW = 7  
 -1.6650E-01 3.3717E-02 5.6935E-01 -1.1020E-01 -5.0237E-01 -1.5215E-02 7.6947E-01 -2.0488E-01 -1.1942E 00 3.7417E-02  
 5.1267E-01 -2.7071E-02 -6.6767E-01 -7.1535E-01 -6.0552E-03 9.5370E-02 1.9640E-02 -1.8095E-02 -1.3271E-02 9.1976E-02  
 -2.9148E-01 -1.8810E-03 1.1203E-01 -4.2155E-03 -6.0552E-03 9.5370E-02 1.9640E-02 -1.8095E-02 -1.3271E-02 9.1976E-02  
 -1.2887E-02 2.2894E-02 -2.9860E-02 -4.4628E-02 -2.1361E-02 1.7765E-02 -2.3074E-02 6.3610E-02 5.4701E-02 -1.2492E-01  
 -4.4193F-02 5.4227E-02 -3.3024E-03 6.2540E-03 7.6334E-03 -1.2225E-02 -5.5486E-03 4.9704E-03  
  
 ROW = 8  
 4.0641E-02 -1.3001E-02 8.1880E-02 -7.1241E-02 -1.2468E-01 -1.9969E-02 -3.8985E-01 -2.3115E-02 8.8648E-01 5.3001E-01  
 -3.5881E-01 -5.4214E-01 -4.0868E-01 -2.9429E-02 1.0651E 00 5.0673E-01 -5.2510E-01 -5.7549E-01 -1.1475E-01 -6.2871E-03  
 2.6668E-01 3.2246E-02 -1.2509E-01 -1.2429E-01 -1.3093E-01 -6.7354E-02 5.3541E-03 4.8114E-02 -6.1503E-03 -9.5861E-02 8.9283E-03 4.8061E-02  
 -1.7227E-02 3.1088E-02 3.7361E-02 -6.0762E-02 -2.6417E-02 2.5516E-02 -4.4061E-02 8.1882E-02 9.5210E-02 -1.5962E-01  
 -6.7531F-02 6.7371E-02 -4.8670E-03 8.6555E-03 1.0547E-02 -1.6851E-02 -7.4004E-03 7.0273E-03  
  
 ROW = 9  
 9.7246E-02 -4.2524E-02 -1.9521E-01 -1.1123E-02 1.1153E-01 8.6897E-03 -6.1485E-01 1.7086E-01 1.3039E 00 1.2241E-01  
 -6.145E-01 -2.8234E-01 -5.4211E-01 1.1642E-01 1.2134E 00 1.6256E-01 -6.1564E-01 -3.0508E-01 -1.6388E-01 3.4369E-02  
 3.4791E-01 3.2246E-02 -1.7397E-01 -6.7354E-02 5.3541E-03 4.8114E-02 -6.1503E-03 -9.5861E-02 8.9283E-03 4.8061E-02  
 -1.4930E-03 1.3870E-02 -1.4947E-02 -2.8227E-02 4.4000E-03 1.6268E-02 1.3698E-02 3.5697E-02 -2.5912E-02 -7.1971E-02  
 4.6547E-03 3.9418E-02 1.9856E-03 3.9771E-03 -3.8437E-03 -8.0333E-03 1.0167E-03 4.5225E-03  
  
 ROW = 10  
 -2.6524E 00 7.0128E-01 5.5980E 00 2.0655E-01 -3.8124E 00 -2.4568E-01 9.8196E 00 -2.3479E 00 -1.6476E 01 4.6466E-01  
 7.8615E 00 1.1595E-01 -5.9086E 00 1.3315E 00 9.4614E 00 -4.7121E-01 -4.6751E 00 1.6912E-01 1.8141E 00 -3.3437E-01  
 -2.8663E 00 -2.9261E-02 1.1966E 00 1.2483E-02 1.5584E-01 6.5422E-02 -3.1052E-01 -1.4407E-01 1.3641E-01 1.1244E-01  
 1.3094E-02 6.3152E-03 -1.9950E-02 -1.3083E-02 6.2079E-03 6.9325E-03 8.3521E-02 2.8001E-02 -1.6420E-01 -6.4170E-02  
 7.3783E-02 5.2369E-02 1.0073E-02 1.5810E-03 -2.0545E-02 -4.1642E-03 9.8864E-03 4.8998E-03  
  
 ROW = 11  
 1.0163F 00 -4.2876E-02 -3.1727E 00 -2.5649E 00 1.3594E 00 2.7617E 00 -3.9969E 00 -6.8642E-02 8.7215E 00 5.1784E 00  
 -3.3293E 00 -5.2429E 00 2.5491E 00 9.1388E-02 -5.4657E 00 -3.0109E 00 2.2073E 00 2.9856E 00 -8.1743E-01 -5.7137E-02  
 1.6870E 00 9.0856E-01 -6.9317E-01 -8.5240E-01 -1.3000E 00 -4.9778E-01 -1.1662E-01 -5.9988E-03 2.3601E-01 2.1945E-02  
 -1.3256E-02 2.4839E-03 2.6875E-02 -4.5455E-03 -1.4262E-02 -8.3073E-04 -3.0449E-01 1.45176E-02 6.0761E-02 6.7708E-03  
 -3.4055E-02 -7.08625E-03 -3.6079E-03 1.1555E-03 2.0844E-03 1.3663E-02 -3.3806E-03 -1.8144E-03 -3.7035E-03 -2.5449E-04  
  
 ROW = 12  
 1.8546E 00 -6.7302E-01 -5.3695E 00 -8.3089E-01 3.4056E 00 1.5153E 00 -7.1817E 00 2.0610E 00 1.4954E 01 9.5450E-01  
 -7.3994E 00 -2.7891E 00 4.4804E 00 -1.1369E 00 -9.1433E 00 -5.3659E 01 4.5062E 00 1.5021E 00 -1.4133E 00 2.6118E-01  
 2.7683E 00 3.1602E-01 1.3000E 00 -4.9778E-01 -1.1662E-01 -5.9988E-03 2.3601E-01 2.1945E-02 -1.1622E-01 -4.1794E-02  
 -1.0821E-02 5.1266E-03 1.8313E-02 -9.841E-03 -9.0170E-03 4.0158E-03 -6.0761E-02 6.7708E-03 1.2179E-01 -8.3683E-03  
 -6.1632E-02 -1.1117E-02 -6.5526E-03 2.0844E-03 1.3663E-02 -3.3806E-03 -1.8144E-03 -3.7035E-03 -2.5449E-04  
  
 ROW = 13  
 1.7857E-01 4.4691E-01 -1.0520E 00 1.8833E 00 2.3132E 00 -1.3544E 00 -1.0367E 00 -7.4173E 00 2.0610E 00 1.4954E 01 9.5450E-01  
 -5.9522E 00 5.2294E-01 4.1441E-01 -1.9336E-01 -2.6532E 00 1.9051E 00 2.5945E 00 -1.1130E 00 -3.0552E-01 1.4919E-01  
 9.1660E-01 -5.4716E-01 -1.0200E-01 2.4275E-01 -7.0346E-01 -2.4962E-01 3.5532E 00 4.2461E-01 -2.6370E 00 1.3966E 00  
 -5.7426E-01 6.6749E-01 -2.2474E 00 -1.1693E 00 2.2283E 00 1.7897E 00 -2.7489E-01 1.7743E-01 2.2635E 00 2.8597E-01  
 -1.8214E-01 -9.5100E-01 -2.5101E-02 8.9459E-02 -2.6651E-01 -1.5631E-01 2.6776E-01 2.4320E-01  
  
 ROW = 14  
 -1.5773F-01 3.9393E-01 4.427E 00 -1.8001E-01 -2.4001F-01 -3.8944E-01 1.1982F-01 1.6001E 00 -6.1914E-01 1.6420E 00

-1.1131E-01	-4.6001E-01	-1.1358E-02	4.4783E-01	1.7055E-01	-1.2622E-01	1.6513E-01	-2.0709E-01	-3.4354E-02	-2.9495E-02
-1.2098E-01	-1.0143E-02	-1.1569E-01	4.8450E-03	-1.7572E-00	-2.1752E-00	3.1172E-00	4.3031E-00	-9.8920E-01	-2.3906E-00
1.4619E-00	2.7322E-00	-2.4610E-00	-5.3569E-00	5.3850E-01	2.8180E-00	-1.2195E-00	-1.6138E-00	2.1495E-00	3.1841E-00
-6.5610E-01	-1.7501E-00	1.8842E-01	3.7423E-01	-3.1521E-01	-7.3303E-01	6.2806E-02	3.8209E-01		
ROW = 15									
2.330E-01	1.0459E-02	3.5430E-01	-8.2326E-01	-7.3511E-01	2.5575E-01	-2.4883E-01	1.0812E-00	2.3778E-01	-2.0294E-01
7.4677E-01	-2.6350E-01	-8.2678E-02	3.0534E-02	7.6065E-01	-4.4822E-01	-8.1087E-01	1.1625E-01	7.0002E-02	-6.2878E-02
-2.4713E-01	2.2729E-01	2.2602E-01	-1.2034E-01	-4.0709E-00	"1.9686E-00	5.6896E-00	4.2544E-00	-1.3865E-00	-2.0883E-00
4.9120E-00	2.4273E-00	-6.5087E-00	-5.2361E-00	1.3404E-00	2.4114E-00	-3.1180E-00	-1.4743E-00	4.3025E-00	3.1697E-00
-1.0201E-00	-1.5337E-00	6.7209E-01	3.4352E-01	-8.8593E-01	-7.3752E-01	1.7753E-01	3.3802E-01		
ROW = 16									
9.7333E-02	2.2462E-01	-3.4806E-01	9.8626E-01	1.0235E-00	-8.1356E-01	-5.3085E-01	-7.8688E-01	1.6966E-00	-7.80559E-02
-1.9509E-00	1.2056E-01	6.3154E-01	-2.4563E-01	-1.8733E-00	1.5485E-00	1.6112E-01	-9.4112E-01	-2.2785E-01	1.1922E-01
-1.469E-01	-3.7432E-01	-5.1547E-01	1.5288E-01	-5.1547E-01	1.2525E-02	2.1073F-01	-1.5074E-03	-1.5074E-00	-5.6790E-01
-3.3845E-01	3.4188E-01	-8.8022E-01	5.9718E-01	9.6673E-01	8.7055E-01	-3.7102E-01	-1.0411E-01	1.9143E-00	1.6667E-01
-1.4341E-00	-6.7715E-01	-7.0655E-02	6.4402E-02	-1.4852E-01	-1.1216E-01	1.8261E-01	1.6509E-01		
ROW = 17									
-7.1194E-02	4.5570E-01	7.119337-01	-4.5889E-02	-1.6659E-01	-2.8609E-01	6.9566E-02	-8.2534E-01	-3.3685E-01	8.9457E-01
-2.3274E-02	-2.7791E-01	4.5835E-02	2.0888E-01	-6.0743E-03	7.3301E-02	1.8542E-01	-2.1861E-01	-2.0477E-02	-1.1571E-02
9.5919E-02	-1.00339E-02	-8.2646E-02	-8.7813E-04	-1.0011E-00	-1.0673E-00	1.7993E-00	2.1194E-00	-6.1685E-01	-1.2469E-00
6.2024E-01	1.3098E-00	-1.0313E-00	-2.5632E-00	1.8675E-01	1.3325E-00	-9.6720E-01	-1.1717E-00	1.7183E-00	2.3172E-00
-5.5246E-01	-1.2906E-00	1.1902E-01	2.5699E-01	-1.9619E-01	-5.0269E-01	3.3244E-02	2.6027E-01		
ROW = 18									
1.1262E-01	1.2606E-02	2.0605E-01	-3.8246E-01	-3.7128E-01	7.9246E-02	-1.0312E-01	5.5329E-01	6.6641E-02	-1.3647E-01
4.1748E-01	-1.2534E-01	-6.5067E-02	-2.0404E-03	4.6562E-01	"2.3948E-01	-4.8971E-01	6.3434E-02	5.8799E-02	-5.2135E-02
-1.9668E-01	1.6883E-01	1.7141E-01	-8.5746E-02	-2.1141E-00	-9.9160E-01	2.9954E-00	2.1699E-00	-7.6393E-01	-1.7033E-00
2.3791E-00	1.1575E-00	-3.1041E-00	-2.5002E-00	6.0937E-01	1.1281E-00	-2.2351E-00	-1.0780E-00	3.1327E-00	2.3264E-00
-7.7085E-01	-1.1459E-00	4.6326E-01	2.3567E-01	-6.0539E-01	-5.0624E-01	1.1781E-01	2.2957E-01		
ROW = 19									
1.5093E-03	-1.1922E-02	4.5605E-01	3.4232E-02	-3.9528E-01	-2.6150E-01	8.2738E-02	8.5741E-02	-2.6159E-01	2.88446E-01
4.8144E-01	-2.3693E-01	4.3693E-01	-1.0996E-01	-1.4362E-01	2.9467E-01	-2.0569E-01	4.3632E-01	4.2728E-02	-5.22663E-02
-1.3337E-01	1.7684E-01	1.2743E-01	-1.0820E-01	-3.2841E-01	2.1896E-01	3.0891E-01	-4.0568E-01	-3.3626E-01	1.90117E-01
-1.6544E-01	-3.7669E-02	6.5306E-01	6.4867E-02	-4.5678E-01	-2.1007E-01	-4.2502E-01	5.1893E-02	1.0546E-00	-9.6670E-02
-6.26244E-01	-1.37886E-01	-5.5997E-02	-5.1731E-03	1.7932E-01	8.9204E-03	-1.1765E-01	4.64493E-02		
ROW = 20									
2.8456E-02	-3.1318E-02	9.7848E-02	1.1288E-01	-6.5034E-02	-1.7461E-01	4.6570E-03	1.2507E-01	3.3422E-02	-6.2725E-02
4.5943E-02	-2.7200E-02	6.0363E-02	-6.0682E-02	-7.8910E-02	2.4620E-01	9.6039E-02	-2.0155E-01	9.4359E-03	-1.1761E-02
-2.8944E-02	4.6447E-02	2.9913E-02	-2.0764E-02	-1.4921E-01	1.5596E-01	3.0987E-01	-2.9251E-01	1.0461E-01	
-3.0778E-01	-3.5723E-01	5.8707E-01	7.0764E-01	-1.0022E-01	3.9691E-01	4.3225E-01	-3.1177E-01	7.9379E-01	6.2739E-01
-3.0940E-01	-3.85336E-01	8.1635E-02	-8.4609E-02	1.4631E-01	1.6814E-01	-5.0362E-02	-9.5888E-02		
ROW = 21									
2.1094E-03	2.7608E-04	-4.2907E-02	7.0633E-02	-2.8342E-00	1.0809E-01	1.4891E-00	2.0342E-00	-4.6395E-00	8.7J24E-01
-5.4993E-03	-2.5858E-03	1.0830E-02	-2.2646E-02	-5.0574E-02	2.3019E-02	3.0946E-02	5.6664E-03	-2.8181E-03	8.7843E-04
9.5123E-03	-1.3771E-02	-1.1878E-02	1.0292E-02	1.2018E-01	8.5450E-02	-8.3595E-02	-1.8216E-01	-4.2024E-02	5.3432E-02
-6.4401E-01	-3.3122E-01	9.1404E-01	7.1164E-01	-2.2806E-01	-3.5603E-01	-5.6358E-01	-3.2240E-01	5.0629E-01	9.0742E-01
-2.7446E-01	-3.7856E-01	-1.4970E-01	-8.2816E-02	2.1903E-01	1.7645E-01	-5.7943E-02	-9.1120E-02		
ROW = 22									
-1.4971E-01	-3.3313E-01	2.1942E-00	-1.1568E-00	-2.8342E-00	1.0809E-01	1.4891E-00	2.0342E-00	-4.6395E-00	8.7J24E-01
5.5259E-00	-8.8569E-01	-1.4621E-00	5.2583E-01	5.0192E-00	-2.2506E-00	-4.3521E-00	5.5710E-01	5.2841E-01	-3.1825E-01
-1.6434E-00	1.0566E-00	1.3536E-00	-4.969UE-01	-4.5688E-01	5.7966E-01	-1.4181E-00	-1.0177E-01	1.5736E-00	1.4150E-01
3.3445E-01	-4.2883E-01	3.6441E-00	1.4351E-00	-3.4690E-00	-2.5117E-00	-3.3233E-01	5.0629E-01	6.2739E-01	-1.7975E-00
1.8196E-00	1.5032E-00	6.0601E-02	-1.1313E-01	5.9792E-01	1.9173E-01	-5.7775E-01	-3.9409E-01		

ROW	COLUMN	DATA
1	1	-2.0565F-01 -8.0686E-02 -8.5198E-01 6.-701.4 -01 1.-0143E 00 6.-1760E-02 3.-5416E-01 -2.-0140E 00 -6.-3374E-01 9.-6288E-01
1	24	-2.0565F-01 -8.0686E-02 -8.5198E-01 6.-701.4 -01 1.-0143E 00 6.-1760E-02 3.-5416E-01 -2.-0140E 00 -6.-3374E-01 9.-6288E-01
2	1	-2.2106E-01 -8.0338E-01 -8.5549E-01 5.2321E-01 1.-4384E-01 *1.-8912E-01 -6.-9677E-02 2.-0586E 00 7.-1144E-01 -2.-0528E 00
2	2	-2.0426E-01 5.0656E-01 -6.9858E-02 -4.3042E-01 3.-7710E-01 2.-9247E-01 -5.-1211E-01 -1.-1207E-01 6.-5240E-02 -2.-2074E-03
2	3	-2.0952E-01 1.0733E-01 1.9236E-01 -6.1027E-02 1.-0829E 00 2.-1325E 00 -1.-8180E 00 -4.-1789E 00 3.-6955E-01 -2.1871E 00
2	4	-2.2537E 00 -3.9255E 00 3.8527E 00 7.7132E 00 -9.2740E-01 -4.0917E 00 1.-2467E 00 2.-2909E 00 -2.1150E 00 -4.4919E 00
2	5	-4.7698F-01 2.3679E 00 -3.7839E-01 -6.4956E-01 6.-4562E-01 1.2751E 00 -1.-5660E-01 -6.-7641E-01
3	1	-8.0338E-01 -8.5549E-01 -6.9858E-02 -4.3042E-01 3.-7710E-01 2.-9247E-01 -5.-1211E-01 -1.-1207E-01 6.-5240E-02 -2.-2074E-03
3	2	-6.0922E-01 -3.7313E-01 -3.9255E-01 -7.0922E-01 -9.0949E-02 -1.-4694E 00 5.-8700E-01 1.-3421E-02 -1.-1450E-01 1.-0617E-01
3	3	-4.3844E-01 -3.5404E 00 9.5566E 00 7.6371E 00 -1.-9965E 00 -3.-5318E 09 4.-2137E 00 1.-9569E 00 -5.-6791E 00 -4.2444E 00
3	4	-7.1850E-01 -1.9863E 00 -1.2013E 00 -6.1491E-01 1.5998E 00 -1.3195E 00 -3.-3212E-01 -6.1197E-01
4	1	-2.0565F-01 -8.0686E-02 -8.5198E-01 6.-701.4 -01 1.-0143E 00 6.-1760E-02 3.-5416E-01 -2.-0140E 00 -6.-3374E-01 9.-6288E-01
4	2	-6.0922E-01 -3.7313E-01 -3.9255E-01 -7.0922E-01 -9.0949E-02 -1.-4694E 00 5.-8700E-01 1.-3421E-02 -1.-1450E-01 1.-0617E-01
4	3	-4.3844E-01 -3.5404E 00 9.5566E 00 7.6371E 00 -1.-9965E 00 -3.-5318E 09 4.-2137E 00 1.-9569E 00 -5.-6791E 00 -4.2444E 00
4	4	-7.1850E-01 -1.9863E 00 -1.2013E 00 -6.1491E-01 1.5998E 00 -1.3195E 00 -3.-3212E-01 -6.1197E-01

SAMPLE PROBLEM TYPICAL MISSILE  
FLUTTER ANALYSIS BY A COLLOCATION METHOD USING AERODYNAMIC INFLUENCE COEFFICIENTS

NSUR = 1 AERO = 1 NRIGID = 2 NFUS = 0 NDENS = 1 MODES OUT = 6 NRELH = 1 NPUNCH = 0  
B (REF) = 0.1330000E 01 K = 0.1000000E 11  
SURFACE H S EXTERNAL STORES SIZE  
1 0.13300000E 01 0.33330000E 01 0  
) C0LUMN 1 COLUMN 2 COLUMN  
1 0.10000000E-09 0.10000000E-09  
2 0.









RIGID BODY MODAL MATRIX  
SURFACE 1,  
34 CONTROL POINTS

COLUMN 1, COLUMN 2

	COLUMN 1	COLUMN 2
1	0.1000000E+01	0.1500000E+02
2	0.1000000E+01	0.1500000E+02
3	0.1000000E+01	0.2500000E+02
4	0.1000000E+01	0.3500000E+02
5	0.1000000E+01	0.1500000E+02
6	0.1000000E+01	0.5500000E+02
7	0.1000000E+01	0.6500000E+02
8	0.1000000E+01	0.7500000E+02
9	0.1000000E+01	0.8500000E+02
10	0.1000000E+01	0.1000000E+03
11	0.1000000E+01	0.1600000E+02
12	0.1000000E+01	0.5565500E+02
13	0.1000000E+01	0.6572800E+02
14	0.1000000E+01	0.4813006E+02
15	0.1000000E+01	0.5667100E+02
16	0.1000000E+01	0.6562500E+02
17	0.1000200E+01	0.5020500E+02
18	0.1000300E+01	0.5812500E+02
19	0.1000600E+01	0.6000400E+02
20	0.1000900E+01	0.5229000E+02
21	0.1000000E+01	0.5937500E+02
22	0.1000000E+01	0.6645500E+02
23	0.1000000E+01	0.8166000E+02
24	0.1000000E+01	0.8500000E+02
25	0.1000000E+01	0.9166000E+02
26	0.1000000E+01	0.8166000E+02
27	0.1000000E+01	0.8500000E+02
28	0.1000000E+01	0.9166000E+02
29	0.1000000E+01	0.8166000E+02
30	0.1000000E+01	0.8500000E+02
31	0.1000000E+01	0.9166000E+02
32	0.1000000E+01	0.8166000E+02
33	0.1000000E+01	0.8500000E+02
34	0.1000000E+01	0.9166000E+02

FLEXIBILITY MATRIX  
SURFACE 1,  
34 CONTROL POINTS

	COLUMN 1	COLUMN 2	SURFACE 34	COLUMN 4	COLUMN 5	COLUMN 6
	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6
1	0.83072800E-04	0.43437700E-04	0.21776100E-04	0.72571800E-05	0.	-0.16051300E-05
2	0.43437700E-04	0.25228400E-04	0.13565100E-04	0.47587800E-05	0.	-0.10700900E-05
3	0.21776100E-04	0.13565100E-04	0.80910499E-05	0.Jn031700E-05	0.	-0.71339600E-06
4	0.72571800E-05	0.47587800E-05	0.30931700E-05	0.14275400E-05	0.	-0.35669900E-06
5	0.	0.	0.	0.	0.	0.
6	-0.16051300E-05	-0.10700900E-05	-0.71339600E-06	-0.35669900E-06	0.	0.23788200E-06
7	0.	0.	0.	0.	0.	0.
8	0.21394400E-05	0.14263000E-05	0.95087200E-06	0.4753800E-06	0.	-0.35673300E-06
9	0.42786800E-05	0.28526000E-05	0.19017400E-05	0.95087500E-06	0.	-0.71346400E-06
10	0.74880300E-05	0.49920400E-05	0.32804000E-05	0.16603000E-05	0.	-0.12485600E-05
11	-0.41524300E-06	-0.27683000E-06	-0.18455400E-06	0.92277300E-07	0.	-0.34715600E-07
12	-0.29355000E-06	-0.19567100E-06	-0.13044866E-06	-0.65223300E-07	0.	0.25248900E-07
13	0.21381800E-07	0.13341300E-07	0.88912200E-08	0.44471300E-08	0.	-0.54193700E-08
14	-0.11099600E-05	-0.73997800E-06	-0.49332100E-06	-0.24666100E-06	0.	0.92011700E-07
15	-0.59161100E-06	-0.39449000E-06	-0.26294000E-06	-0.13147100E-06	0.	0.44196600E-07
16	-0.41861100E-07	-0.27907600E-07	-0.18605100E-07	-0.90910100E-08	0.	-0.70910100E-08
17	-0.16806200E-05	-0.11204200E-05	-0.74695000E-06	-0.37347700E-06	0.	0.13739600E-06
18	-0.92676000E-06	-0.61784300E-06	-0.41189700E-06	-0.20549000E-06	0.	0.67301800E-07
19	-0.17889700E-06	-0.11926500E-06	-0.79510700E-07	-0.39755500E-07	0.	-0.21219000E-06
20	-0.21875000E-05	-0.14583000E-05	-0.97233200E-06	-0.56181800E-06	0.	0.17670900E-06
21	-0.12731000E-05	-0.84873900E-06	-0.56582900E-06	-0.26291600E-06	0.	0.91699700E-07
22	-0.35927000E-06	-0.23951400E-06	-0.15967700E-06	-0.79387000E-07	0.	0.67598700E-08
23	0.35643600E-05	0.23762600E-05	0.15841800E-05	0.79205400E-06	0.	-0.59432700E-06
24	0.42789500E-05	0.28526500E-05	0.19017700E-05	0.95059100E-05	0.	-0.71347600E-05
25	0.57038000E-05	0.38025500E-05	0.25350500E-05	0.12675300E-05	0.	-0.95105700E-06
26	0.35650900E-05	0.23767400E-05	0.15845000E-05	0.79255300E-06	0.	-0.59444500E-06
27	0.42796700E-05	0.28531300E-05	0.19020900E-05	0.95105200E-06	0.	-0.71359500E-06
28	0.57046200E-05	0.38031000E-05	0.29354100E-05	0.12671700E-05	0.	-0.95119100E-06
29	0.35660800E-05	0.23764000E-05	0.15849400E-05	0.79245300E-06	0.	-0.59460900E-06
30	0.42806800E-05	0.28538000E-05	0.19025400E-05	0.95125000E-05	0.	-0.71376200E-06
31	0.57055900E-05	0.38037400E-05	0.25558400E-05	0.12679300E-05	0.	-0.95135200E-06
32	0.356671800E-05	0.23781300E-05	0.15854300E-05	0.79272700E-06	0.	-0.59479100E-06
33	0.42817600E-05	0.28545200E-05	0.19032000E-05	0.95151500E-06	0.	-0.71394000E-06
34	0.57066300E-05	0.38044400E-05	0.25363000E-05	0.12681600E-05	0.	-0.95152400E-06
					0.	0.25248900E-07
					0.	COLUMN 12
1	0.	0.	0.	0.	0.	0.
2	0.	0.	0.	0.	0.	0.
3	0.	0.	0.	0.	0.	0.
4	0.	0.	0.	0.	0.	0.
5	0.	0.	0.	0.	0.	0.
6	0.	0.	0.	0.	0.	0.
7	0.	0.	0.	0.	0.	0.
8	0.	0.	0.	0.	0.	0.
9	0.	0.	0.	0.	0.	0.
10	0.	0.	0.	0.	0.	0.
11	0.	0.	0.	0.	0.	0.
12	0.	0.	0.	0.	0.	0.
13	0.	0.	0.	0.	0.	0.
14	0.	0.	0.	0.	0.	0.
15	0.	0.	0.	0.	0.	0.
16	0.	0.	0.	0.	0.	0.
17	0.	0.	0.	0.	0.	0.
18	0.	0.	0.	0.	0.	0.



6	-0.21219000E-08	0.17670900E-06	0.916999700E-07	0.67598700E-08	-0.59432700E-06	-0.71347600E-06
7	0.	0.	0.	0.	0.	0.
8	0.48245400E-07	-0.22071300F-06	-0.83681600E-07	0.522799800E-07	0.25371300E-05	0.30935000E-05
9	0.96486700E-07	-0.44142600E-06	-0.16776300E-06	0.10560000E-06	0.60257000E-05	0.76155900E-05
10	0.16885200E-06	-0.77249500E-06	-0.29358500E-06	0.18479900E-06	0.11257000E-04	0.14755900E-04
11	0.52835900E-05	0.11093200E-05	0.90873700E-04	0.90873700E-04	0.761100E-05	0.93170200E-07
12	0.102500E-04	0.13114000E-04	0.12684300E-04	0.12362200E-04	0.59595600E-07	0.71543200E-07
13	0.2252200E-04	0.492552700E-05	0.19988600E-04	0.30727100E-04	0.28706300E-07	0.34461200E-07
14	0.41822800E-04	0.24743700E-03	0.15961600E-03	0.71792400E-04	0.20223000E-06	0.24277200E-06
15	0.1093200E-03	0.16894900E-03	0.15582700E-03	0.16166500E-03	0.754956200E-07	0.90631500E-07
16	0.1669900E-03	0.72656100E-04	0.15582700E-03	0.23918500E-03	0.62753900E-07	0.75334600E-07
17	0.97486100E-04	0.57875800E-03	0.37131300E-03	0.14460000E-03	0.293339400E-06	0.35221300E-06
18	0.2301600E-03	0.37354900E-03	0.36895500E-03	0.36384500E-03	0.80375000E-07	0.96488400E-07
19	0.3663900E-03	0.16575200E-03	0.36308500E-03	0.56146700E-03	0.80375000E-07	0.96488400E-07
20	0.1655200E-03	0.952666800E-03	0.612666300E-03	0.27337400E-03	0.36771500E-16	0.44143300E-06
21	0.3618500E-03	0.612666300E-03	0.60940200E-03	0.59980400E-03	0.13974900E-06	0.16776600E-06
22	0.561146700E-03	0.27337400E-03	0.59980400E-03	0.30666600E-03	0.87966200E-07	0.10560100E-06
23	0.8017500E-07	-0.36771500E-06	-0.13974900E-06	0.87966200E-07	0.23743800E-04	0.11234700E-04
24	0.96488400E-07	-0.44143700E-06	-0.16776600E-06	0.10560100E-06	0.11234700E-04	0.12824700E-04
25	0.12861800E-06	-0.58842700E-06	-0.22353000E-06	0.14076600E-06	0.46898500E-05	0.15994700E-04
26	0.80391800E-07	-0.367778700E-06	-0.13977600E-06	0.67984800E-07	0.36326300E-04	0.26867500E-04
27	0.965105300E-07	-0.41450600E-06	-0.16779300E-06	0.14078700E-06	0.28457500E-04	0.31628500E-04
28	0.12863700E-06	-0.58850900E-06	-0.22356100E-06	0.14727800E-06	0.75275200E-05	0.42503600E-04
29	0.80414800E-17	-0.36788700E-06	-0.13961300E-06	0.88010400E-07	0.49935700E-04	0.42503600E-04
30	0.96529700E-07	-0.44160800E-06	-0.16783100E-06	0.10566600E-06	0.40146900E-04	0.44093600E-04
31	0.12861800E-06	-0.58866700E-06	-0.22369700E-06	0.14082100E-06	0.20464700E-04	0.24264600E-04
32	0.80440400E-07	-0.36799900E-06	-0.13989500E-06	0.88008700E-07	0.63574900E-04	0.58141000E-04
33	0.96553700E-07	-0.44171700E-06	-0.16787100E-06	0.10567400E-06	0.53546000E-04	0.59730700E-04
34	0.12868400E-06	-0.58871200E-06	-0.22373600E-06	0.14083900E-06	0.33562400E-04	0.62900500E-04
	COLUMN 25	COLUMN 26	COLUMN 27	COLUMN 28	COLUMN 29	COLUMN 30
1	0.57038000E-05	0.35650900E-05	0.42796700E-05	0.57046200E-05	0.35660800E-05	0.42806800E-05
2	0.38025500E-05	0.23767400E-05	0.26531300E-05	0.38031300E-05	0.23774000E-05	0.28538000E-05
3	0.25350500E-05	0.15845000E-05	0.19020900E-05	0.25354100E-05	0.15849400E-05	0.19025400E-05
4	0.12675300E-05	0.79225300E-06	0.95105200E-06	0.12677100E-05	0.79247300E-06	0.95127500E-06
5	0.	0.	0.	0.	0.	0.
6	-0.95105700E-06	-0.59444500E-06	-0.71359500E-06	-0.95119100E-06	-0.59460900E-06	-0.71376200E-06
7	0.	0.	0.	0.	0.	0.
8	0.42029000E-05	0.25376500E-05	0.30940200E-05	0.42034900E-05	0.25383700E-05	0.40947500E-05
9	0.10785800E-04	0.60269800E-05	0.76168800E-05	0.10787300E-04	0.60287400E-05	0.76186700E-05
10	0.21730900E-04	0.11260300E-04	0.14758400E-04	0.21737300E-04	0.11263700E-04	0.14761800E-04
11	-0.12462400E-06	-0.77626400E-07	-0.931985800E-07	-0.12423300E-06	-0.77647800E-07	-0.93207500E-07
12	-0.25366400E-07	-0.59607500E-07	-0.71555100E-07	-0.25379800E-07	-0.59623900E-07	-0.71571800E-07
13	-0.45936500E-07	-0.28712100E-07	-0.34467100E-07	-0.45943100E-07	-0.28720100E-07	-0.34475200E-07
14	-0.32361300E-06	-0.202270100E-06	-0.24281200E-06	-0.32365800E-06	-0.20232500E-06	-0.24286900E-06
15	-0.12041100E-06	-0.55104900E-07	-0.64643300E-07	-0.12041100E-06	-0.55314100E-07	-0.96667100E-07
16	0.10042000E-06	0.62766800E-07	0.75347600E-07	0.10043500E-06	0.62784600E-07	0.75365700E-07
17	-0.46949700E-06	-0.29345200E-06	-0.39227100E-06	-0.46956200E-06	-0.29353200E-06	-0.35235300E-06
18	-0.16864400E-06	-0.10540800E-06	-0.12653600E-06	-0.16866700E-06	-0.10543600E-06	-0.12656500E-06
19	0.12861800E-06	0.40391800E-07	0.96505300E-07	0.12866700E-07	0.801414800E-07	0.96528700E-07
20	-0.58842700E-06	-0.36778700E-06	-0.44150600E-06	-0.58850900E-06	-0.36788700E-06	-0.44160800E-06
21	-0.22363100E-06	-0.13977600E-06	-0.16779300E-06	-0.22366100E-06	-0.13981300E-06	-0.16783100E-06
22	0.14076600E-06	0.37984800E-07	0.10562200E-06	0.14078700E-06	0.88010400E-07	0.10564600E-06
23	-0.46898500E-05	0.36362600E-05	0.26563300E-05	-0.46898500E-05	-0.49935700E-05	-0.40146900E-05
24	0.159941700E-04	0.26867500E-04	0.31628500E-04	0.159941700E-04	0.42503600E-04	0.44093600E-04
25	0.92992400E-04	-0.0918800E-05	0.30420400E-04	-0.95104000E-04	0.65334300E-05	0.37983400E-04
26	-0.110191400E-05	-0.4027700E-03	0.21467600E-03	-0.21467600E-03	0.62871400E-03	0.42285900E-03
27	0.30420400E-04	0.21467600E-04	0.22524400E-03	0.21662200E-03	0.42681300E-03	0.43011800E-03
28	0.95104700E-04	-0.219671600E-04	0.27524400E-03	-0.73997900E-03	0.54321800E-03	0.46787600E-03

29	0.65334330E-05	0.62871400E-03	0.42681300E-03	0.54321800E-04
30	0.37983400E-04	0.42585900E-03	0.4301800E-03	0.46787600E-03
31	0.100590UE-03	0.19526600E-04	0.4434500E-03	0.1291900E-02
32	0.14282200E-04	0.88898800E-03	0.6400500E-03	0.121259100E-03
33	0.44782800E-04	0.624919200E-03	0.6400500E-03	0.1291900E-02
34	0.10565500E-03	0.97523500E-04	0.64319400E-03	0.2011200E-03
1	0.57055900E-05	0.15671800E-05	0.42817600E-05	0.14888900E-03
2	0.38037400E-05	0.23781300E-05	0.28545200E-05	0.17772800E-02
3	0.25358400E-05	0.15854300E-05	0.12030200E-05	0.25363000E-05
4	0.12679300E-05	0.79271800E-06	0.95151500E-06	0.12681600E-05
5	0.	0.	0.	0.
6	-0.95135200E-06	-0.59479100E-06	-0.7139400E-06	-0.95152400E-06
7	0.	0.	0.	0.
8	0.-42041900E-05	0.25391600E-05	0.30955300E-05	0.42049400E-05
9	0.-10789000E-04	0.60306900E-05	0.7629800E-05	0.10790800E-04
10	0.-21737100E-04	0.11267500E-04	0.14765500E-04	0.21740700E-04
11	-0.-12423400E-06	-0.77671600E-07	-0.93230800E-07	-0.12425600E-06
12	-0.-95395900E-07	-0.59642100E-07	-0.71589700E-07	-0.95413200E-07
13	0.-45950900E-07	0.28729000E-07	0.34484000E-07	0.45959400E-07
14	-0.-32371300E-06	-0.24227100E-06	-0.24227100E-06	-0.32377100E-06
15	-0.-12084800E-06	-0.75554100E-07	-0.-90689400E-07	-0.12086900E-06
16	-0.-10045200E-06	-0.62804300E-07	-0.75385100E-07	-0.10047100E-06
17	-0.-46964100E-06	-0.29362100E-06	-0.35224400E-06	-0.46972500E-06
18	-0.-16869500E-06	-0.10546800E-06	-0.12655000E-06	-0.16872500E-06
19	0.-12866000E-06	0.89440000E-07	0.96551500E-07	0.12868400E-06
20	-0.-58866700E-06	-0.36799200E-06	-0.4411700E-06	-0.-588671200E-06
21	-0.-22369700E-06	-0.13985500E-06	-0.16797100E-06	-0.-223673600E-06
22	0.-14081200E-06	0.-88038700E-07	0.16797100E-06	0.14083900E-06
23	0.-20464700E-04	0.63571600E-04	0.53546000E-04	0.33562100E-04
24	0.-47264400E-04	0.-58141000E-04	0.5973700E-04	0.6290500E-04
25	0.10059000E-03	0.14292200E-04	0.44782800E-04	0.10565500E-03
26	0.19526600E-04	0.38898800E-03	0.6249200E-03	0.97523500E-04
27	0.-45344500E-03	0.64006500E-03	0.64699100E-03	0.64319400E-03
28	0.-12910000E-02	0.-18843300E-03	0.71836900E-03	0.17772890E-02
29	0.-21259700E-03	0.-20101200E-02	0.14888900E-02	0.4476100E-03
30	0.97833700E-03	0.15159900E-02	0.1516300E-02	0.15115000E-02
31	0.25n53900E-02	0.52507100E-03	0.15n5300E-03	0.36269700E-02
32	0.52507100E-03	0.-32866300E-02	0.2507800E-02	0.96514300E-03
33	0.15593300E-02	0.25097810F-02	0.25096100E-02	0.24979500E-02
34	0.36269700E-02	0.-26514300E-03	0.-24979200E-02	0.355671200E-02

SURFACE 1. WEIGHTING MATRIX  
NO WEIGHTING MATRIX







30	-0.6533400E-01	-0.1746100E 00	0.4657000E-02	0.12507400E 00	0.33422000E-01	0.62724500E-01
31	0.7261300E-01	-0.4270900E-01	0.55441600E-01	-0.18009100E 00	-0.1346800E 00	0.12313400E 00
32	-0.2834200E-01	-0.10809100E 00	0.14891000E-01	-0.29342100E 01	-0.46394800E 01	0.87324400E-01
33	0.1383900E 00	-0.18911600E 00	-0.69677300E-01	0.20585800E-01	0.71144200E 00	-0.20528100E 01
34	0.1013500E 01	0.61759600E-01	0.35466700E 00	-0.2014000E 01	-0.63373800E 04	0.96267700E 00
	31	COLUMN 31	COLUMN 32	COLUMN 33	COLUMN 34	COLUMN 35
	32					COLUMN 36
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16	-0.24693000E+01	-0.864689700E+00	0.55620400E+00	-0.97583800E+01	0.13673900E+01
17	0.17594000E+00	-0.15353300E-01	0.19434700E+00	-0.38972600E+01	-0.18810100E-02
18	-0.52509700E+00	-0.57500000E+00	-0.11475000E+00	-0.62871300E+02	0.26667600E+00
19	-0.61563900E+00	0.16386200E+00	-0.16386200E+00	0.34378900E+01	0.32445000E+00
20	-0.46751100E+01	0.16911800E+00	0.18141000E+01	-0.33437200E+00	-0.28663400E+01
21	0.22027200E+01	0.29855200E+01	-0.81742900E+00	-0.57036800E+01	-0.29260600E+01
22	0.45619000E+01	0.15020900E+01	-0.14143400E+01	0.26116100E+00	0.90553000E+00
23	0.29554000E+01	-0.11790000E+01	-0.30519000E+00	0.14919400E+00	0.97060300E+00
24	0.16512600E+00	0.20708600E+00	-0.34353500E+01	-0.29494700E+01	0.12096100E+00
25	-0.81067300E+00	0.11625200E+00	0.70002300E+01	-0.62877600E+01	-0.24713000E+00
26	0.18111500E+01	-0.94112000E+00	-0.22784800E+00	0.11302200E+00	0.71466800E+00
27	0.18542400E+00	-0.21861100E+00	-0.28477000E+00	-0.11570600E+01	0.95949200E+01
28	-0.48971360E+00	0.63333700E+01	-0.58708000E+01	0.52135000E+01	-0.54716000E+00
29	0.29467000E+00	-0.53635400E+00	0.42728200E+01	-0.52663400E+01	0.13337200E+00
30	0.96039000E+01	-0.20155400E+00	0.94358600E+00	-0.11781200E+01	0.28394000E+01
31	0.36694600E+01	0.566663700E+02	-0.28181200E+02	0.87843000E+03	0.95123200E+02
32	-0.43321100E+01	0.75709900E+00	0.52640800E+00	-0.31825100E+00	-0.16434100E+01
33	-0.51220500E+00	-0.1120206800E+00	0.65239800E+01	-0.22074000E+02	0.20952100E+00
34	0.13431300E+01	-0.63421300E+01	-0.13449700E+01	0.10617300E+00	0.43884000E+00
		COLUMN 44	COLUMN 45	COLUMN 46	
1	1	0.	0.	0.	
2	2	0.	0.	0.	
3	3	0.	0.	0.	
4	4	0.	0.	0.	
5	5	0.	0.	0.	
6	6	0.	0.	0.	
7	7	0.	0.	0.	
8	8	0.	0.	0.	
9	9	0.	0.	0.	
10	10	0.	0.	0.	
11	11	-0.71212700E+00	-0.866658300E+03	-0.29557300E+00	
12	0	-0.34811600E+00	-0.48334500E+00	-0.77726000E+01	
13	0	0.72163900E+00	0.28566300E+00	0.19261100E+00	
14	-0	-0.49881200E+00	-0.59816600E+02	-0.15963900E+00	
15	0	0.23030500E+00	0.31899900E+00	-0.70944300E+01	
16	0	-0.49173400E+00	0.10471300E+00	0.19565600E+01	
17	0	0.11203200E+00	-0.42154500E+02	-0.60551700E+02	
18	-0	-0.12428500E+00	-0.13092700E+00	-0.56266900E+01	
19	-0	-0.17397000E+00	-0.67353800E+01	0.93541500E+02	
20	0	-0.11966100E+01	0.14862700E+01	-0.65222000E+01	
21	0	-0.69317500E+00	-0.85406300E+00	-0.523348400E+01	
22	-0	-0.12999800E+01	-0.49778200E+00	-0.11662200E+01	
23	-0	-0.80200300E+00	-0.24274900E+00	-0.70346200E+00	
24	-0	-0.15692000E+00	0.48450100E+02	-0.17572100E+01	
25	0	-0.22602000E+00	-0.12033700E+00	-0.40708900E+01	
26	-0	-0.57452500E+00	0.15287500E+00	-0.51547300E+00	
27	-0	-0.82645500E+01	-0.87812000E+03	-0.16010800E+01	
28	0	-0.17140700E+00	-0.85746500E+01	-0.21141000E+01	
29	0	-0.12742600E+00	-0.10619800E+00	-0.32640800E+00	
30	0	-0.29912800E+01	-0.30583600E+01	-0.14920700E+00	
31	-0	-0.1878800E+01	0.10292300E+01	0.12018100E+00	
32	0	-0.13536000E+01	-0.49690300E+00	-0.57963800E+00	
33	-0	-0.19236300E+00	-0.613026800E+01	0.19629300E+01	
34	-0	-0.38662400F+00	0.19724100F+00	-0.17857000E+01	
		COLUMN 49	COLUMN 50	COLUMN 51	
1	1	0.	0.	0.	
2	2	0.	0.	0.	

0.

0.

0.

0.

0.



26	-0.96672500E+00	0.87054800E+00	-0.37102500E+00	-0.10411000E+00	0.19143300E+01	0.16667200E+00
27	0.-18675300E+00	0.13324900E+01	-0.96720200E+00	-0.11717200E+01	0.17183500E+01	0.23171500E+01
28	0.-60937300E+00	0.-11261300E+01	-0.22352000E+01	-0.-10782700E+01	0.-31327400E+01	0.-23284300E+01
29	-0.-45677600E+00	-0.42502500E+00	0.-51873200E+01	0.-10546100E+01	0.-96670300E+01	0.-966738500E+00
30	-0.-16022100E+00	-0.-39691400E+00	-0.43225000E+00	-0.-31276600E+00	0.-79378000E+00	0.-62738500E+00
31	-0.-22805800E+00	-0.-35602700E+00	-0.-58358100E+00	-0.-32240000E+00	0.-90742500E+00	0.-68666200E+00
32	-0.-34689900E+01	-0.-29147600E+01	-0.-33233100E+00	0.-58628600E+00	-0.-17952000E+01	-0.-10267000E+01
33	-0.-92739600E+00	-0.-40917100E+01	0.-12466200E+01	0.-22908800E+01	-0.-21149900E+01	-0.-44918700E+01
34	-0.-19964600E+01	-0.-35318400E+01	0.-42137200E+01	0.-19569400E+01	-0.-56799900E+01	-0.-32443900E+01

COLUMN 61 COLUMN 62 COLUMN 63 COLUMN 64 COLUMN 65 COLUMN 66

1	0.	0.	0.	0.	0.	0.
2	0.	0.	0.	0.	0.	0.
3	0.	0.	0.	0.	0.	0.
4	0.	0.	0.	0.	0.	0.
5	0.	0.	0.	0.	0.	0.
6	0.	0.	0.	0.	0.	0.
7	0.	0.	0.	0.	0.	0.
8	0.	0.	0.	0.	0.	0.
9	0.	0.	0.	0.	0.	0.
10	0.	0.	0.	0.	0.	0.
11	-0.16693500E+00	0.33885100E+01	-0.11588900E+01	0.91471000E+02	0.24095700E+01	-0.17106600E+01
12	-0.81974900E+01	0.14924400E+00	-0.16009900E+02	0.15618400E+01	0.45074500E+02	-0.38836700E+01
13	0.-84517600E+01	0.-30000000E+01	0.-95017500E+02	0.-50122000E+02	-0.-18626000E+01	0.-18769000E+01
14	-0.-11674900E+00	-0.45701100E+01	-0.-85713200E+02	0.-61661600E+02	-0.18201800E+01	-0.15450000E+01
15	-0.-88377300E+01	0.-11659200E+00	-0.-33314700E+02	0.-12934300E+01	-0.77723400E+02	-0.-22438800E+01
16	-0.77932000E+01	0.-52369000E+01	-0.10072500E+01	0.-15889800E+02	-0.20544600E+02	-0.-41642900E+02
17	-0.-44193300E+01	-0.-54226700E+01	-0.-33024300E+02	0.-65066100E+02	-0.76333500E+02	-0.-12225000E+01
18	-0.-67531300E+01	-0.-67376600E+01	-0.-48676400E+02	0.-86556100E+02	-0.10546300E+01	-0.-16055900E+01
19	-0.-465475600E+02	-0.-39176000E+01	-0.-19856600E+02	-0.-39771500E+02	-0.-38467800E+02	-0.-60336400E+02
20	-0.-73783200E+01	-0.-52369000E+01	-0.10072500E+01	0.-15889800E+02	-0.-20544600E+01	-0.-41642900E+02
21	-0.-34054600E+01	-0.-78824800E+02	-0.-36079200E+02	0.-11550300E+02	-0.71911000E+02	-0.-16143200E+02
22	-0.-61631900E+01	-0.-11173000E+01	-0.-65526100E+02	0.-20846000E+02	-0.13661100E+01	-0.-33666900E+02
23	-0.-18205400E+01	-0.-95092000E+00	-0.-72299700E+01	0.-874423200E+00	-0.-315251200E+00	-0.-73362700E+00
24	-0.-65609300E+00	-0.-17500000E+01	-0.18842400E+00	0.-34352400E+00	-0.-68592600E+00	-0.-73751800E+00
25	-0.-19206000E+01	-0.-15386500E+01	-0.-67202700E+00	0.-64402300E+01	-0.-24852400E+00	-0.-11216100E+00
26	-0.-1.134100E+01	-0.-67715200E+00	-0.-70655300E+01	0.-11906000E+01	-0.-25699200E+00	-0.-19619000E+00
27	-0.-55242000E+00	-0.-12966000E+01	-0.-11458700E+01	0.-46326300E+01	0.-23567200E+00	-0.-60539600E+00
28	-0.-77644900E+00	-0.-11458700E+01	-0.-13786200E+00	-0.-55996800E+01	-0.-51737000E+02	-0.-17932600E+00
29	-0.-62824300E+00	-0.-13786200E+00	-0.-38536400E+00	-0.-81635100E+01	-0.-84689500E+01	-0.-14637000E+00
30	-0.-39939600E+00	-0.-38536400E+00	-0.-38536400E+00	-0.-81635100E+01	-0.-84689500E+01	-0.-14637000E+00
31	-0.-27446500E+00	-0.-37856200E+00	-0.-14970400E+00	-0.-82816200E+01	-0.-21935000E+00	-0.-17645200E+00
32	-0.-18195900E+01	-0.-350351500E+01	-0.-60608800E+01	-0.-11312600E+01	-0.-59791900E+00	-0.-19173100E+00
33	-0.-47698400E+00	-0.-23679100E+01	-0.-37839000E+00	-0.-64956100E+00	-0.-64956100E+00	-0.-12751000E+01
34	-0.-12621390E+01	-0.-19865130E+01	-0.-12012600E+01	-0.-61491300E+01	-0.-15998000E+01	-0.-13195000E+01

COLUMN 67 COLUMN 68 COLUMN 69

1	0.	0.	0.
2	0.	0.	0.
3	0.	0.	0.
4	0.	0.	0.
5	0.	0.	0.
6	0.	0.	0.
7	0.	0.	0.
8	0.	0.	0.
9	0.	0.	0.
10	0.	0.	0.
11	-0.-141461200E+01	0.-1.15646000E+02	0.
12	0.-60435900E+02	0.-1.47665000E+01	0.
13	0.-79719899E+02	0.-77021600E+02	0.
14	-0.-11129700E+01	0.-52387300E+02	0.

15	-0.70421700E-02	0.11642260E-01
16	0.35205200E-02	0.66237100E-02
17	-0.55560560E-02	0.49704000E-02
18	-0.74039000E-02	0.70272900E-02
19	0.10167200E-02	0.45224600E-02
20	0.98864200E-02	0.48998000E-02
21	-0.37034900E-02	-0.39285700E-04
22	-0.73160200E-02	-0.25449000E-03
23	0.28775900E-00	0.24049500E-00
24	-0.62806400E-01	0.38289300E-00
25	0.17752800E-00	0.33801500E-00
26	0.18261400E-00	0.16581600E-00
27	0.33244389E-01	0.26268000E-00
28	0.17867800E-00	0.22957000E-00
29	-0.11765100E-00	-0.46392500E-01
30	-0.50382100E-01	-0.95877900E-01
31	-0.37943200E-01	-0.91119900E-01
32	-0.57775100E-00	-0.33948870E-00
33	-0.15660400E-00	-0.67648600E-00
34	-0.33212000E-00	-0.11967000E-00

## OUTPUT DATA

## FLUTTER ANALYSIS BY A COLLOCATION METHOD, USING AERODYNAMIC INFLUENCE COEFFICIENTS.

DENSITY = 0.2378000E-02 REDUCED VELOCITY = 0.1000000E 01

## 2 RIGID BODY DEGREES OF FREEDOM

## EIGENVALUE

## ITERATIONS S.P. U.P. ATKINS S.P. R.P.

MODE	EIGENVALUE	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6
1	0.17131410E-02	-0.7322634E-63		25	0	4	0
2	0.157342E-02	0.2550746E-05		14	0	3	0
3	0.73801308E-03	-0.1475008E-03		39	0	3	0
4	0.5627548E-03	-0.2686156E-03		17	0	3	0
5	0.3208921E-03	-0.8352808E-04		30	0	4	0
6	0.3626017E-03	0.4706694E-06		18	0	1	0

## EIGENVECTORS

1	0.37643184E-02	0.90765241E-02	-0.1306326E 00	-0.17352749E 00	-0.21184392E-02	0.26283327E-02
2	0.19566667E-02	0.29771531E-02	-0.4331979E-01	-0.57166711E-01	-0.19439889E-03	0.4957239E-03
3	0.79181301E-03	0.17373488E-03	0.46627465E-02	0.6471594E-02	-0.67433479E-03	-0.7229894E-03
4	0.31848380E-03	-0.16189E-02	0.36423231E-01	0.46953620E-01	0.92923979E-03	0.13252424E-02
5	0.13219182E-02	-0.23693806E-02	0.4645105E-01	0.63552799E-01	0.64995364E-03	-0.14494983E-02
6	-0.21969870E-02	-0.21788696E-02	0.4511541E-01	0.6115253E-01	0.50124068E-04	-0.13986483E-02
7	-0.2948243E-02	-0.10329762E-02	0.2739456E-01	0.3975663E-01	-0.34613223E-03	-0.93641223E-03
8	-0.393336945E-02	0.35423904E-02	-0.176972E-01	-0.2355494E-02	-0.161672235E-02	-0.3783438E-03
9	-0.5160556E-02	0.35423904E-02	-0.3024727E-01	-0.45665127E-01	-0.26603667E-02	-0.28876897E-03
10	-0.51691147E-02	0.79435493E-02	-0.10312494E-00	-0.12928819E-00	-0.50164371E-02	0.136531685E-02
11	-0.14536092E-02	-0.31906559E-02	-0.50773181E-01	-0.65362626E-01	-0.92093510E-03	0.24627588E-02
12	-0.22198841E-02	-0.26426640E-02	0.41386E-01	0.53456639E-01	-0.19181078E-02	-0.19181078E-02
13	-0.28237313E-02	-0.18346710E-02	-0.29866990E-01	0.4147667E-01	-0.42863891E-02	0.36461925E-02
14	-0.20512034E-02	-0.93737831E-02	0.7265594E-01	0.7372944E-01	-0.139135E-01	0.27736265E-01
15	-0.19561059E-02	-0.86473640E-02	-0.60781275E-01	0.6592867E-01	-0.20862874E-01	0.25605594E-01
16	-0.16772631E-02	-0.70791439E-02	-0.4524537E-01	-0.56523628E-01	-0.2816621E-01	0.33799398E-01
17	-0.31302892E-02	-0.17253523E-01	-0.10164463E-01	0.8478937E-01	-0.27030114E-01	0.63706813E-01
18	-0.152818668E-02	-0.15943354E-01	0.86086488E-01	0.8222768E-01	-0.45992822E-01	0.7192097E-01
19	-0.51591930E-03	-0.14995830E-01	0.75915013E-01	0.86677708E-01	-0.64144784E-01	0.7745844E-01
20	-0.44211907E-02	-0.25379461E-01	0.1317539E-00	0.66696229E-01	-0.4513494E-01	0.1283377E-01
21	-0.9698137E-03	-0.23961320E-01	0.11414587E-00	0.18101119E-00	-0.74949007E-01	0.1453976E-01
22	0.24460211E-02	-0.22564264E-01	0.96701087E-01	0.1522770E-00	-0.1477070E-00	0.12631485E-00
23	0.16786163E-01	0.24610741E-02	-0.39462011E-02	-0.2472121E-01	0.188491179E-01	-0.45095018E-03
24	0.16421816E-01	0.71617629E-02	-0.1867044E-02	-0.4319221E-01	0.167194544E-01	-0.61184656E-03
25	0.69122601E-02	0.16207645E-01	-0.56213495E-01	-0.73586680E-01	0.10534559E-01	0.11066333E-02
26	0.27405729E-01	-0.18581306E-01	-0.26669872E-00	-0.26273974E-01	0.2719471E-01	0.10405294E-01

27	0.23279708E_00	0.47209725E_01	0.18687098E_00	-0.20641377E_01	0.24293449E_00	-0.19276797E_01
28	0.16143659E_00	-0.18146866E_00	0.43751792E_01	-0.76191586E_02	0.29242910E_00	0.886413B9E_01
29	0.62018514E_00	-0.17005901E_01	0.61905639E_00	-0.15774926E_01	0.61740246E_00	0.92239224E_02
30	0.54078035E_00	-0.10627347E_01	0.47756580E_00	0.87178860E_02	0.56165864E_00	0.51602833E_01
31	0.38014052E_00	-0.35120614E_00	0.19321891E_00	0.57069071E_01	0.44759098E_00	0.17294512E_00
32	0.10000000E_01	0.12542395E_00	0.10000000E_01	0.10000000E_01	0.10000000E_01	0.10000000E_01
33	0.87357339E_00	0.17244078E_00	0.79020295E_00	0.30811136E_01	0.90711855E_00	0.91033076E_01
34	0.62677912E_00	0.51668167E_00	0.37065954E_00	0.11666092E_00	0.72057229E_00	0.27305778E_00
	COLUMN 7	COLUMN 8	COLUMN 9	COLUMN 10	COLUMN 11	COLUMN 12
1	-0.92542652E_03	0.93245277E_03	0.18032191E_02	-0.58704626E_03	-0.24300373E_00	-0.11772136E_00
2	0.31636750E_04	0.26611171E_03	-0.36221644E_03	-0.11646464E_03	0.24894566E_01	0.43441653E_01
3	0.43284813E_03	0.96721044E_04	0.11493591E_02	0.14231952E_03	0.15177272E_00	0.73441653E_01
4	0.48705526E_03	-0.31195582E_03	0.10844431E_02	-0.29033586E_03	0.13384469E_01	0.65015894E_01
5	0.24329271E_03	-0.35432235E_03	-0.62061236E_03	0.27244350E_03	0.39891448E_01	0.19834431E_01
6	-0.16577689E_03	-0.23503363E_03	0.17372986E_03	0.66336593E_04	0.60631729E_01	0.28551957E_01
7	-0.67991192E_03	-0.19960543E_04	0.45244077E_03	-0.98017666E_04	0.11749778E_00	0.58951522E_01
8	0.12574409E_02	0.46278486E_03	0.90909150E_04	0.97438795E_04	0.95320178E_01	0.45005755E_01
9	-0.19643947E_02	0.91924066E_03	-0.10718600E_02	0.53680204E_03	0.38655633E_02	0.39523040E_02
10	-0.37411872E_02	0.15383473E_02	-0.46030034E_02	0.12741641E_02	0.27471270E_00	0.13530793E_00
11	-0.11417451E_03	-0.71545129E_03	0.50542279E_02	0.67172742E_02	0.34318146E_01	0.15199346E_01
12	-0.29938732E_03	-0.40470186E_03	0.20948219E_02	-0.1973077E_02	0.43860261E_01	0.2574105E_01
13	-0.43834097E_03	-0.29019538E_03	0.80556752E_02	-0.13118460E_01	0.10513174E_00	0.63007494E_01
14	-0.37532443E_03	-0.30230092E_02	0.42979485E_01	0.47946120E_01	0.47928373E_01	0.98205066E_02
15	0.52160286E_03	-0.29283103E_02	0.55411647E_01	0.26865470E_01	0.24943135E_01	0.46997291E_01
16	0.13793444E_02	-0.26565667E_02	0.63717150E_01	-0.98244665E_01	0.31200336E_02	0.99882360E_01
17	-0.61452366E_03	-0.62126255E_02	-0.1017100E_00	0.1053964E_00	0.72358344E_01	0.79171836E_02
18	0.17227052E_02	-0.62434917E_02	0.12723265E_00	-0.58349059E_01	0.1199347E_00	0.67376494E_01
19	-0.45228205E_02	-0.63414010E_02	0.15393344E_00	-0.22202783E_00	0.21471769E_00	0.11886873E_00
20	-0.10386913E_02	-0.95996534E_02	0.16521012E_00	0.16316879E_00	0.10139676E_00	0.2531662E_02
21	0.32155529E_02	-0.98975118E_02	0.21267500E_00	-0.92305514E_01	0.23375105E_00	0.84449886E_01
22	0.74579062E_02	-0.10196999E_01	0.26031677E_00	0.15399351E_00	0.36606946E_00	0.17611597E_00
23	0.319174281E_01	-0.28179130E_02	0.22431317E_01	-0.12036367E_02	-0.14713462E_01	0.23840340E_01
24	0.19738147E_01	-0.11183304E_02	0.19771766E_01	-0.2509537E_02	-0.27566769E_01	0.17715723E_02
25	0.12680110E_01	-0.87209297E_03	0.61055722E_02	0.60185481E_03	0.10695511E_00	0.42035179E_01
26	0.26672868E_00	0.17823547E_01	0.28883789E_00	0.21929004E_01	0.28896227E_00	0.56214876E_02
27	0.24994493E_00	-0.31519438E_01	0.23519369E_00	-0.45374707E_01	0.24532537E_00	0.53379141E_01
28	0.23429437E_00	-0.13254088E_00	0.14473836E_00	-0.18331305E_00	0.17523168E_00	0.15391572E_00
29	0.61254309E_00	0.15702000E_01	0.63305524E_00	0.19863356E_01	0.64782121E_00	0.93937626E_03
30	0.57997972E_00	-0.82248185E_01	0.53780810E_00	-0.11630895E_00	0.53220937E_00	0.12311935E_00
31	0.51189152E_00	-0.27741983E_00	0.31497292E_00	-0.38671160E_00	0.29900779E_00	0.36946983E_03
32	0.10000000E_01	0.	0.10000000E_01	0.	0.10000000E_01	0.
33	0.93997215E_00	0.14268887E_00	0.86207719E_00	-0.20056483E_00	0.82006164E_00	0.19983037E_00
34	0.818666782E_00	-0.42791934E_00	0.58511161E_00	-0.600026634E_00	0.46106953E_00	0.59772949E_00

## CHECK EIGENVALUES AND EIGENVECTORS

)	0.17131411E-02	-0.72322613E-03	0.15157343E-02	0.25352235E-05	0.73081400E-03	-0.14774981E-05
)	0.56287563E-03	-0.28860134E-03	0.32888941E-03	-0.83572584E-04	0.30281033E-03	0.7084308E-06
)	COLUMN 1	COLUMN 2	COLUMN 3	COLUMN 4	COLUMN 5	COLUMN 6

	1	2	3	4	5	6
)	1.0 -0.37643246E-02	0.80705214E-02	-0.13083320E 0	"0.17352760E 0	-0.2118130E-02	0.262835903E-02
)	2.0 -0.19566688E-02	0.29771524E-02	-0.43331964E-01	-0.52166746E-01	-0.19431128E-03	0.40958779E-03
)	3.0 -0.79181268E-03	0.17373510E-03	-0.36741592E-02	0.6353012E-01	-0.72228993E-03	-0.72228993E-03
)	4.0 -0.3166567E-03	-0.16344179E-03	0.34332277E-01	0.46853012E-01	0.92815922E-03	-0.13255430E-02
)	5.0 -0.13219208E-02	-0.23693795E-02	0.68451333E-01	0.63556696E-01	0.64085568E-03	-0.14495436E-02
)	6.0 -0.21969895E-02	-0.21788688E-02	0.4511490E-01	0.61157243E-01	0.50030750E-04	-0.13086947E-02
)	7.0 -0.28948261E-02	-0.10320759E-02	0.271673924E-01	0.39157658E-01	-0.73467962E-03	-0.93643329E-03
)	8.0 -0.34236948E-02	-0.45860236E-02	0.11769996E-02	0.23556921E-02	-0.16167234E-02	-0.38783649E-03
)	9.0 -0.39333042E-02	-0.35423889E-02	0.38212462E-01	-0.45605153E-01	-0.2663028E-02	0.2807369E-03
)	10.0 -0.51691101E-02	0.79435459E-02	-0.10312490E 0	-0.12928087E 0	-0.503162424E-02	0.13653763E-02
)	11.0 -0.14536116E-02	-0.33962676E-02	0.50738124E-01	0.65381245E-01	-0.24626847E-02	0.191816583E-02
)	12.0 -0.22198611E-02	-0.26426594E-02	0.4014094E-01	0.53466227E-01	-0.143664189E-02	0.191816583E-02
)	13.0 -0.28237358E-02	-0.18346710E-02	0.98606959E-01	0.4116681E-01	-0.428645683E-02	0.36461597E-02
)	14.0 -0.20512051E-02	-0.93737778E-02	0.72615539E-01	0.7329641E-01	-0.11394277E-01	0.27736205E-01
)	15.0 -0.6660598E-02	-0.8673603E-02	0.67128E-01	0.662995E-01	-0.32560526E-01	0.33799345E-01
)	16.0 -0.18728535E-02	-0.70791391E-02	0.46214592E-01	0.5622032E-01	-0.2831672E-01	0.33799345E-01
)	17.0 -0.31302900E-02	-0.17253515E-01	0.10164457E 0	0.84109743E-01	-0.27030316E-01	0.63705970E-01
)	18.0 -0.15281886E-02	-0.15943345E-01	0.86066433E-01	0.82227008E-01	-0.45993001E-01	0.71092009E-01
)	19.0 -0.151591613E-03	-0.14967662E-03	0.75914971E-01	0.86676627E-01	-0.7765723E-01	0.7765723E-01
)	20.0 -0.44211995E-02	-0.25379440E-01	0.13137753E 0	0.9669640E-01	-0.45135617E-01	0.10283366E-01
)	21.0 -0.96908354E-03	-0.23961305E-01	0.11418581E 0	0.101010121E 0	-0.7498923E-01	0.11453961E 0
)	22.0 -0.24460176E-02	-0.22564246E-01	0.96701038E-01	-0.10527724E 0	-0.10477092E 0	0.12631468E 0
)	23.0 -0.16786183E-01	0.2641732E-02	-0.3961909E-02	-0.2741326E-01	0.1891216E-01	-0.5003408E-03
)	24.0 -0.16421817E-01	0.71617020E-02	-0.18660712E-01	-0.43179245E-01	0.16791910E-01	-0.81163412E-03
)	25.0 -0.69122624E-02	0.16207665E-01	-0.56013467E-01	-0.73580721E-01	-0.10535564E-01	0.11186672E-02
)	26.0 -0.27406729E-06	-0.10580301E-01	0.26669874E-00	-0.26273080E-01	-0.27194723E-00	0.10405294E-01
)	27.0 -0.23279770E-06	0.7729719E-01	-0.1867098E-00	-0.20454387E-01	-0.24292498E-00	-0.19278772E-01
)	28.0 -0.16746651E-09	0.181460087E-01	0.43751754E-01	-0.76792125E-02	0.23242986E-00	-0.8041261E-01
)	29.0 -0.62018515E-09	-0.70585892E-01	0.6195642E-00	-0.15777925E-01	0.51740253E-00	0.92799688E-02
)	30.0 -0.54076007E-06	0.10627349E-00	0.47756577E-00	0.87178482E-02	0.56165861E-01	-0.51682801E-01
)	31.0 -0.38014637E-06	0.35120611E-00	0.13218776E-00	-0.57063136E-01	0.44756799E-00	-0.17294589E-01
)	32.0 -0.10000000E-01	0.16024681E-09	0.14900000E-01	0.14353747E-10	0.10000000E-01	0.1492589E-03
)	33.0 -0.87357331E-09	0.17244074E-09	0.9202898E-00	0.38671121E-01	0.90711847E-00	-0.91033105E-01
)	34.0 -0.62077890E-09	0.51668162E-06	0.37065931E-00	0.41646469E-00	0.72057235E-00	-0.27305726E-00
)	COLUMN 7	COLUMN 8	COLUMN 9	COLUMN 10	COLUMN 11	COLUMN 12

	1	2	3	4	5	6	7	8	9	10	11	12
)	-0.92584975E-03	0.93262222E-03	0.186181662E-02	-0.58767274E-03	-0.24390627E-00	-0.11772163E-00	-0.24089233E-01	-0.55895023E-01	-0.14431230E-01	-0.734415210E-01	-0.65915777E-01	-0.19534347E-01
)	2.0 -0.31762364E-04	0.26616180E-03	-0.3681460E-03	-0.11681460E-03	-0.56044307E-01	-0.15777264E-00	-0.7334415210E-01	-0.15777264E-00	-0.53032099E-01	-0.45805668E-01	-0.39521810E-01	-0.13534632E-01
)	3.0 -0.48695703E-03	-0.31201121E-03	-0.10956431E-02	0.29295966E-03	0.133846447E-00	0.51740253E-00	0.389910235E-01	0.399910235E-01	0.66632182E-01	-0.55895023E-01	-0.19534347E-01	-0.27471348E-01
)	4.0 -0.24318340E-03	-0.35442113E-03	-0.46231326E-03	0.27227053E-03	0.399910235E-01	-0.399910235E-01	-0.19534347E-01	-0.399910235E-01	-0.66632182E-01	-0.55895023E-01	-0.19534347E-01	-0.27471348E-01
)	5.0 -0.16587866E-03	-0.23513230E-03	0.17370759E-03	0.66193522E-04	-0.66193522E-04	-0.11774982E-01	-0.55895023E-01	-0.11774982E-01	-0.55895023E-01	-0.55895023E-01	-0.11774982E-01	-0.55895023E-01
)	6.0 -0.67999191E-03	0.49847129E-04	0.4522609E-03	-0.9897932E-04	-0.9897932E-04	-0.21749925E-01	-0.55895023E-01	-0.21749925E-01	-0.55895023E-01	-0.55895023E-01	-0.21749925E-01	-0.55895023E-01
)	7.0 -0.40477574E-03	-0.29026713E-03	0.8458439E-02	0.97637173E-04	-0.97637173E-04	-0.16532182E-01	-0.63007412E-01	-0.16532182E-01	-0.63007412E-01	-0.63007412E-01	-0.16532182E-01	-0.63007412E-01
)	8.0 -0.12574041E-02	0.46278147E-03	0.90966303E-03	-0.1016343E-02	-0.53684196E-03	-0.53684196E-02	-0.474426463E-01	-0.474426463E-01	-0.474426463E-01	-0.474426463E-01	-0.474426463E-01	-0.474426463E-01
)	9.0 -0.19464303E-02	0.919279780E-03	-0.15384384E-02	-0.46233409E-02	0.1274226E-02	0.1274226E-02	-0.34317455E-01	-0.34317455E-01	-0.34317455E-01	-0.34317455E-01	-0.34317455E-01	-0.34317455E-01
)	10.0 -0.37409898E-02	0.71762861E-03	0.23513230E-03	0.2095614E-02	-0.19574110E-02	-0.19574110E-02	-0.43080626E-01	-0.43080626E-01	-0.43080626E-01	-0.43080626E-01	-0.43080626E-01	-0.43080626E-01
)	11.0 -0.11666693E-03	0.49847129E-04	0.4522609E-03	-0.9897932E-04	-0.9897932E-04	-0.21749925E-01	-0.63007412E-01	-0.21749925E-01	-0.63007412E-01	-0.63007412E-01	-0.21749925E-01	-0.63007412E-01
)	12.0 -0.29949489E-03	-0.40477574E-03	0.8458439E-02	0.8458439E-02	-0.1318583E-01	-0.1318583E-01	-0.63007412E-01	-0.63007412E-01	-0.63007412E-01	-0.63007412E-01	-0.63007412E-01	-0.63007412E-01
)	13.0 -0.43841636E-03	-0.29026713E-03	0.30231203E-02	0.42794088E-01	0.474426463E-01	0.474426463E-01	-0.474426463E-01	-0.474426463E-01	-0.474426463E-01	-0.474426463E-01	-0.474426463E-01	-0.474426463E-01
)	14.0 -0.37548815E-03	-0.474426463E-01	0.55476562E-01	0.26632605E-01	-0.24922605E-01	-0.24922605E-01	-0.49973795E-01	-0.49973795E-01	-0.49973795E-01	-0.49973795E-01	-0.49973795E-01	-0.49973795E-01
)	15.0 -0.13279234E-02	-0.265666803E-02	0.63316828E-01	-0.28648382E-01	-0.2127151E-02	0.10563396E-01	0.720577235E-01	0.720577235E-01	-0.31704383E-01	-0.99662379E-01	-0.99662379E-01	-0.791727781E-01
)	16.0 -0.13279234E-02	-0.265666803E-02	0.63316828E-01	-0.2127151E-02	0.10563396E-01	0.10563396E-01	0.720577235E-01	0.720577235E-01	-0.31704383E-01	-0.99662379E-01	-0.99662379E-01	-0.791727781E-01
)	17.0 -0.61478865E-03	-0.62127151E-02	0.10563396E-01	0.10563396E-01	0.10563396E-01	0.10563396E-01	0.720577235E-01	0.720577235E-01	-0.31704383E-01	-0.99662379E-01	-0.99662379E-01	-0.791727781E-01

18	-0.17225058E-02	-0.62436510E-02	-0.12723211E+00	-0.58796207E-01	0.11993883E+00	-0.67376600E-01
19	-0.45226913E-02	-0.63416453E-02	-0.15393032E+00	-0.20780624E+00	0.21971902E+00	-0.19391049E-02
20	-0.10389878E-02	-0.95998119E-02	-0.16520946E+00	-0.16516877E+00	0.10139602E+00	-0.62531884E-02
21	-0.32153178E-02	-0.98977481E-02	-0.21267430E+00	-0.95305719E-01	0.23375024E+00	-0.84449741E-01
22	-0.74576566E-02	-0.10197321E-01	-0.26031623E+00	-0.36359403E+00	0.36660868E+00	-0.17631666E+00
23	-0.19174331E-01	-0.28182762E-03	-0.22431251E-01	-0.14608828E-02	-0.14713282E-01	-0.23840352E-01
24	-0.17974823E-01	-0.19771925E-01	-0.25049087E-02	-0.27565043E-01	-0.127716990E-02	-0.127716990E-02
25	-0.12680259E-01	-0.67201233E-03	-0.61059595E-02	-0.60174196E-03	-0.10695563E+00	-0.42034919E-01
26	-0.26672877E+00	-0.17823589E-01	-0.28843806E+00	-0.21926766E-01	-0.28866246E+00	-0.56215993E-02
27	-0.24994492E+00	-0.31519362E-01	-0.4519382E+00	-0.45374561E-01	-0.24323252E+00	-0.533749213E-01
28	-0.23429433E+00	-0.1325073E+03	-0.14743862E+00	-0.18381261E+00	-0.17501578E+00	-0.19391049E-03
29	-0.61254319E+00	-0.15702065E+01	-0.63305357E+00	-0.19603308E+01	-0.6482142E+00	-0.12311946E+00
30	-0.57997976E+00	-0.82248144E+01	-0.53780814E+00	-0.11630879E+00	-0.5322940E+00	-0.36946987E+00
31	-0.51189137E+00	-0.27741979E+00	-0.34497288E+00	-0.38671102E+00	-0.29900775E+00	-0.18497447E-11
32	-0.10000000E+01	-0.15264014E+08	-0.10000000E+01	-0.68620129E+09	-0.19000000E+01	-0.19983097E+00
33	-0.93997211E+00	-0.14266900E+00	-0.86207720E+01	-0.20056466E+00	-0.82066171E+00	-0.19983097E+00
34	-0.81866740E+00	-0.42791915E+00	-0.58511584E+00	-0.60026543E+00	-0.46106928E+00	-0.59772946E+00

		$1/k_r = 1.00$	DAMPING	VELOCITY (KNOTS)
MODE	OMEGA (CPS)			
1	0.7554/063E 02	-0.42216393E 00	-	0.37464703E 03
2	0.80316100E 02	0.16725060E -02	-	0.39829726E 03
3	0.1150394nE 03	-0.1999281E 00	-	0.57049433E 03
4	0.13179774E 03	-0.51272719E 00	-	0.65368096E 03
5	0.17294732E 03	-0.25566096E 00	-	0.85766673E 03
6	0.17969207E 03	0.15541400E -02	-	0.89111472E 03

	$1/k_r = .666$	DAMPING	VELOCITY (KNOTS)
MODE	OMEGA (CPS)		
1	0.780200381E 02	-0.17405223E -02	0.26491021E 03
2	0.93454299E 02	-0.26253529E 00	0.38865929E 03
3	0.11795594E 03	-0.10306314E 00	0.38950047E 03
4	0.14495615E 03	-0.31769220E 00	0.47875933E 03
5	0.17988994E 03	-0.16137299E -03	0.59386694E 03
6	0.20905074E 03	-0.42464674E 00	0.69944759E 03

	$1/k_r = .500$	DAMPING	VELOCITY (KNOTS)
MODE	OMEGA (CPS)		
1	0.80334597E 02	-0.71199755E -02	0.19919449E 03
2	0.93887853E 02	-0.24626489E 00	0.23280096E 03
3	0.11683951E 03	-0.33438035E -01	0.28971064E 03
4	0.15019281E 03	-0.31623869E 00	0.37241216E 03
5	0.17953335E 03	0.17292386E -02	0.44516301E 03
6	0.21403295E 03	-0.14626305E -01	0.548220227E 03

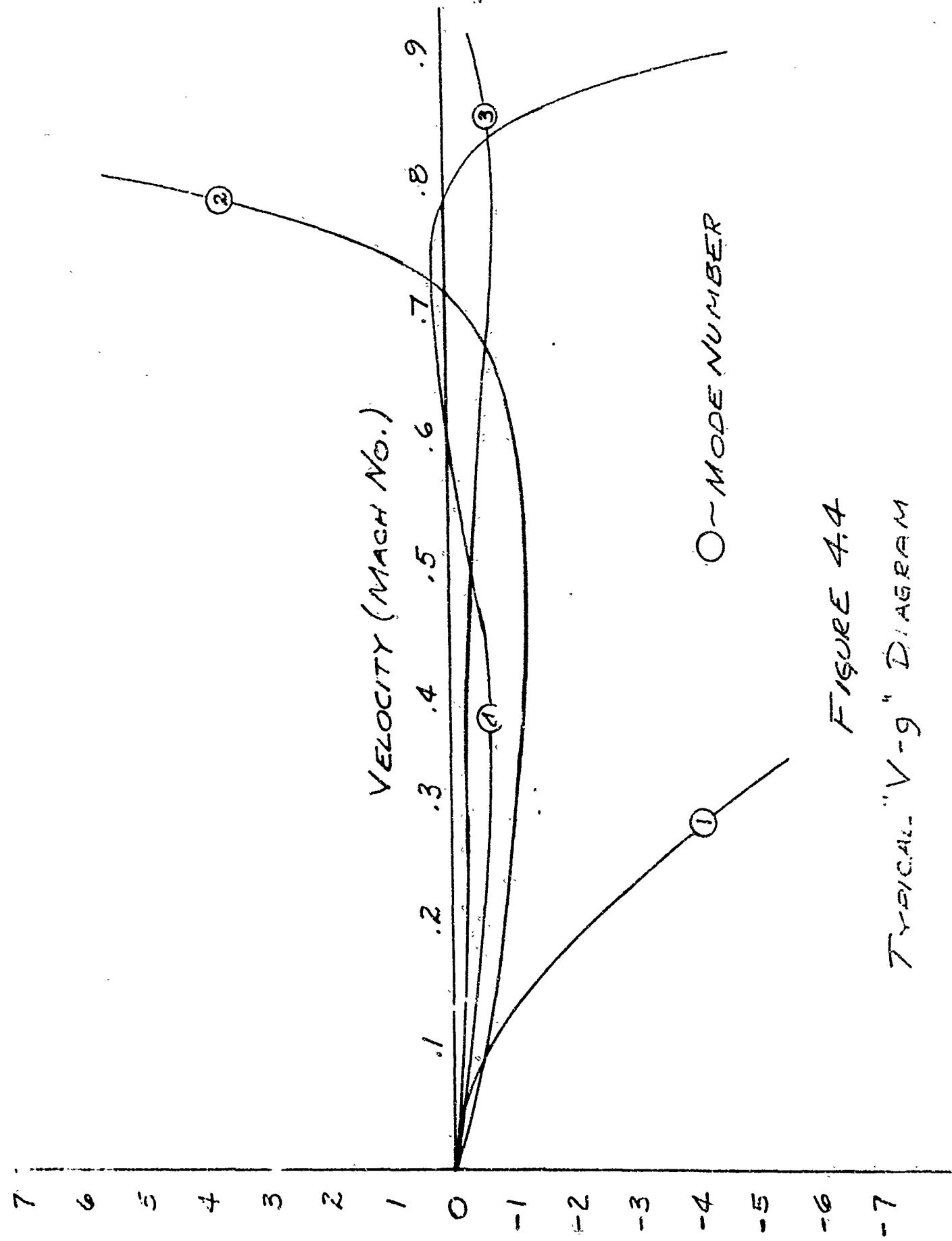


FIGURE 4.4

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13. ABSTRACT

THIS STUDY COVERS THE DEVELOPMENT OF A SET OF COMPUTER PROGRAM TO PERFORM FLUTTER ANALYSIS BY THE COLLOCATION METHOD. WHILE THIS METHOD HAS BEEN KNOWN FOR SOME TIME, ONLY RECENTLY HAVE ADVANCES IN COMPUTER TECHNOLOGY MADE THE METHOD TECHNICALLY AND FINANCIALLY FEASIBLE. THE INGREDIENTS OF A COLLOCATION FLUTTER ANALYSIS ARE 1) A FLEXIBILITY MATRIX, 2) AERODYNAMIC INFLUENCE COEFFICIENT MATRIX, AND 3) AN EIGENVALUE SOLUTION. THIS STUDY IS PRESENTED IN FOUR VOLUMES. VOLUME I CONTAINS A GENERAL PROGRAM DISCUSSION. VOLUME II CONTAINS THE PROGRAM FLURNC WHICH CALCULATES THE FLEXIBILITY MATRIX. VOLUME III CONTAINS A SET OF THREE PROGRAMS TO CALCULATE AERODYNAMIC INFLUENCE COEFFICIENTS FOR SUBSONIC, TRANSONIC, AND SUPERSONIC FLIGHT REGIMES. VOLUME IV CONTAINS THE PROGRAM COFA WHICH SETS UP AND SOLVES THE FLUTTER EIGENVALUE MATRIX.

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FLUTTER						
VIBRATION						
AERODYNAMIC INFLUENCE COEFFICIENTS						

~~UNCLASSIFIED~~

Security Classification